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COMPUTATIONAL FLUID DYNAMICS (CFD) ANALYSIS OF A C-135 AIRCRAFT WITH A SIDE-MOUNTED SPLITTER PLATE (with comparison to wind tunnel data)

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June 1994

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Test Wing at Wright-Patte	erson AFB. The data co	ollected from the a	analysis was provided to the
Test Wing to augment win	d tunnel tests and to p	rovide loading an	d stability information on the
modified aircraft. This wo	rk was a precursor to a	a planned flight te	est program in which a
splitter plate will be tested	l on a Test Wing aircra	ft. Four flight co	nditions were analyzed with a
modified and unmodified C	J-135 aerodynamic mod	lel, and increment	tal stability and aerodynamic
derivatives were determine	ed Additionally press	ure coefficient dat	ta were tabulated for use in a

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### **Foreword**

The subject analysis was performed by Mr Howard Emsley and Mr Ken Wurtzler of the Flight Dynamics Directorate, Wright Laboratory (WL/FIMC) at the request of the Airborne Laser (ABL) SPO at Phillips Lab, Kirtland AFB. Capt John Wissler of the ABL SPO served as the directing authority for this support effort and Dr James Van Kuren provided direction as the resident consultant for the ABL SPO. This analysis was performed to support a planned ABL flight test in which a large splitter plate configuration will be mounted on the side of a C-135 aircraft.

# Acknowledgements

This work was performed with the cooperation of personnel from the Airborne Laser (ABL) SPO at Kirtland AFB, the 4950TW at Wright-Patterson AFB, and Dr James Van Kuren who is serving as an ABL SPO consultant. A special thanks goes to members of Wright Laboratory (WL/FIMC) for their code developing efforts which provided the tools necessary to perform this analysis.

All wind tunnel data used in this report comes from tests conducted at the Trisonic Gasdynamics Facility, Wright Laboratory (WL/FIME) in April 1993. This data was made available by Dr Van Kuren, for comparison with the numerical results.

# Nomenclature

Alpha Angle of Attack

AOA Angle of Attack

Beta Sideslip Angle

CD Coefficient of Drag

CFD Computational Fluid Dynamics

CL Coefficient of Lift

CMpitch Pitching Moment Coefficient

CMyaw Yawing Moment Coefficient

CMroll Rolling Moment Coefficient

Cp Coefficient of Pressure

CPU Central Processing Unit

CY Coefficient of Yaw

deg Degrees (angular)

FIMC Computational Fluid Dynamics Branch, Aeromechanics Division,

Flight Dynamics Directorate

KEAS Knots Equivalent Air Speed

mxx Moment about the X-Axis

myy Moment about the Y-Axis

mzz Moment about the Z-Axis

px Pressure in the X Direction

py Pressure in the Y Direction

pz Pressure in the Z Direction

q Dynamic Pressure

Ve Velocity

WL Wright Laboratory

# 1. Introduction

The splitter plate/optical window configuration (see Figure 1) was designed to provide an environment where optical testing can be performed without the interference of the turbulent boundary layer created by the aircraft's fuselage. The pylon, which is mounted on the side of the aircraft, places the splitter plate at least 12 inches away from the fuselage and outside the fuselage boundary layer. By mounting the splitter plate in this fashion, the optical testing will only have to contend with a regenerated thin boundary layer that starts at the front of the plate.

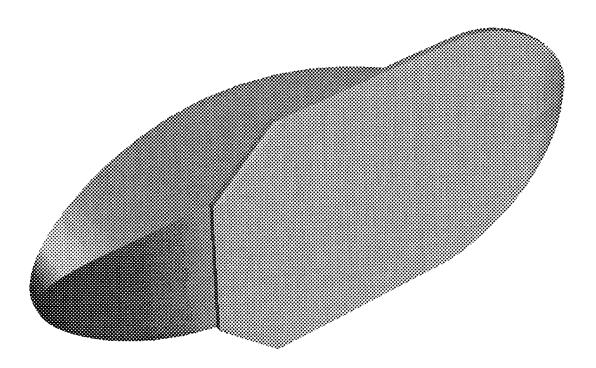


Figure 1: Splitter Plate and Pylon

Concern over the flight characteristics of the modified aircraft as well as concerns about structural loading prompted wind tunnel and Computational Fluid Dynamics (CFD) work to be performed by WL/FIME and WL/FIMC respectively. Structural analysis of the configuration is being performed by the 4950TW/AMDA with pressure data provided by the CFD analysis.

Some results from the wind tunnel testing will be presented in this report, however, complete results are found in Reference 1. For results from the structural analysis, the reader is directed to Mr Kelly Kennedy, 4950TW/AMDA.

The splitter plate/pylon design, geometric measurements of the splitter plate, and its mounting location on the aircraft were provided to WL/FIMC by Dr Jim Van Kuren and Capt Wissler. The test conditions for the CFD analysis (see Table 1) were provided to WL/FIMC by Mr C. E. Cook of 4950TW/AMDA on 1 March 1993 and were confirmed by the ABL SPO in early March.

Table 1: Flight Conditions

Test Pt.	Altitude	Airspeed	Mach No.	AOA	Sideslip	CL**
	(feet)	(KEAS)		(deg)	(deg)	
1	23,800	393	0.95	1.4	4.1	0.21
2	23,800	393	0.95	5.5	4.1	0.525
3	0	240	0.36	14.6	14.5	1.16
4	45,000	189	0.76	3.0	0.0	0.58

\*\*Note: the CL values provided in this table are for a complete
C-135 in trim. The analyses performed do not include
nacelle/pylon contributions and do not include the effects
of rudder, elevator, or flap deflections.

It was explained to WL/FIMC that concern over changes in the flight characteristics of the modified aircraft was the main reason a CFD analysis was requested. Therefore, it was proposed that the total aircraft be modeled (with and without the splitter plate) and analyzed at the four test

points. By proceeding in this fashion, incremental effects of adding the plate/pylon configuration could be determined. For this report the C-135 aircraft without the splitter plate is called the "clean" configuration, and the C-135 aircraft with the plate and pylon mounted on the right side of the fuselage is called the "dirty" configuration.

#### 2. Geometric Issues

From previous work for the 4950TW, WL/FIMC had a suitable model of a C-135 wing/body/tail configuration without engine pylons and nacelles. For the dirty case, the plate/pylon geometry was created with a local CAD/CAM system and added to the existing C-135 geometry. A small support strut which is located under the front of the plate for structural strength was omitted from this analysis due to the increased geometric complications that it would introduce and the minor aerodynamic effects it would produce.

For the dirty configuration, the plate/pylon (location provided by Capt John Wissler) was added to the geometry with the front of the elliptical pylon located at fuselage station 405.35 inches. Based on the provided dimensions, the center of the test window is located at fuselage station 460 inches (see Figure 2). The location of the plate/pylon was driven by aircraft modification limitations.

For this analysis, the pylon was placed with zero angle of attack relative to the fuselage of the aircraft. The minimum distance between the plate and the fuselage was specified as 12 inches by Dr Van Kuren and occurs between the lower half of the plate and the fuselage approximately one quarter of the distance aft from the front of the pylon. The very front of the plate which is dipped towards the fuselage is just under 13 inches away from the fuselage.

For the calculation of the coefficients, the following reference areas, moment reference lengths, and moment reference points were provided by the 4950TW/AMDA:

#### Wing Reference Area = 700700 sq in

Moment x = 1570.00 in Moment x = 846.64 in Reference y = 241.88 in Reference y = 0.00 in Lengths z = 1570.00 in Point z = 200.00 in

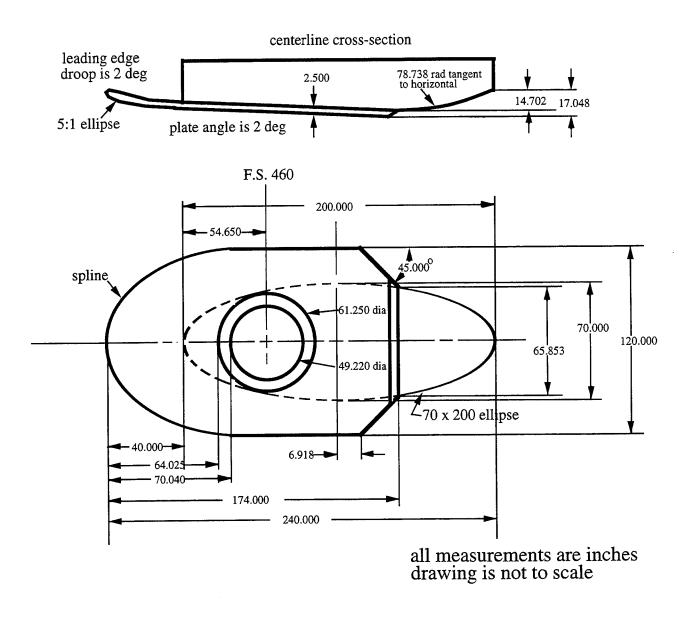


Figure 2: Splitter Plate and Pylon Dimensions

The coordinate system orientation and the positive moment directions are shown below in the figure below.

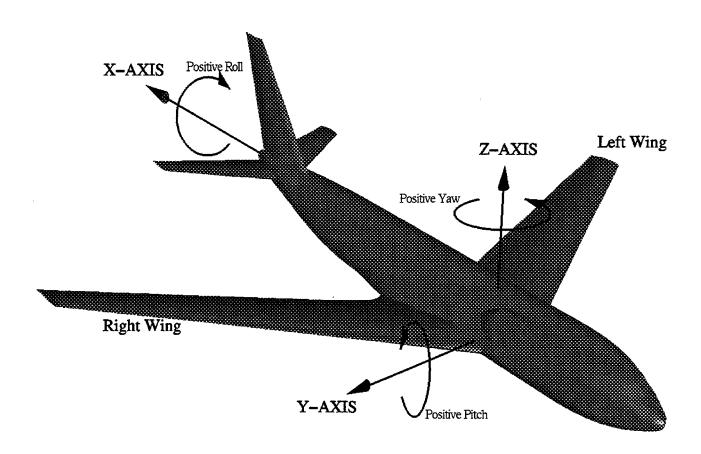


Figure 3: Coordinate System Orientation

# 3. Grid Generation

With the use of in-house grid generation tools (I3G/VIRGO and PLUTO) [2,3], separate computational grids were generated for each configuration (clean and dirty). The grid for the clean configuration consisted of 22 computational blocks with a total of 1,366,720 grid points and the dirty configuration consisted of 33 computational blocks with a total of 1,379,250 grid points. Differences in the size and the number of blocks in each grid is a direct result of geometric complications added by the plate and pylon.

#### 4. Flow Solver

The flow solver used for this analysis, MERCURY, is an in-house Euler code that was developed by Mr William Strang of WL/FIMC [4]. Over the past six years, this code has proven to be very fast, robust, and accurate in calculating flows like those occurring in this analysis. MERCURY is an inviscid flow solver and therefore cannot model the boundary layer that is a concern for this program. It can, however, provide conservative loading information, capture shocks that may occur, and indicate the paths of particles and vortices. This is not a claim that an inviscid solver provides exact measurements in highly viscous regions. Yet, from past experience this solver has produced load information that has proven to be conservative in regions where viscous flows exist. With this in mind, the flight conditions provided for test point #3 contain an extreme beta condition that can produce significant separation on the leeward side of the aircraft. The results in this case should be viewed qualitatively, especially for the dirty configuration where significant separation is expected to occur on the outboard side of the plate. For additional information on the code specifics, the reader is directed to AFWAL-TM-88-217, "MERCURY User's Manual."

Viscous solvers, such as the TEAM and GASP codes, were considered for this problem, but the time constraints and the computational resources available did not allow an analysis of this magnitude.

# 5. Flow Conditions and Computation Time

For this report, a total of nine flow solution "cases" (five clean cases and four dirty cases) have been run. The Kirtland Cray II was used to calculate the solutions with each case requiring approximately 35 MW of internal memory. On average, 6000 iterations per solution were needed to reach acceptable levels of solution convergence (density residual reduced by 4 orders of magnitude). Each clean case took 28 seconds per iteration while each dirty case took 29 seconds/iteration. This translates into 168,000 seconds per solution (46.67 CPU hours) for each clean case and 174,000 seconds per solution (48.33 CPU hours) for each dirty case. The flow conditions for each case are based on the conditions provided by the 4950TW with several modifications. For the clean cases the actual conditions used are found in Table 2, and for the dirty cases the conditions used are found in Table 3.

Table 2: Clean Case Test Conditions

Mach Number	Angle of Attack (Alpha)	Sideslip Angle (Beta)
0.95	5.5 <b>de</b> g	-4.1 deg
0.95	1.4 deg	-4.1 deg
0.87	1.4 deg	-4.1 deg
0.36	14.6 deg	-14.5 deg
0.76	3.0 deg	0.0 deg

Table 3: Dirty Case Test Conditions

Mach Number	Angle of Attack (Alpha)	Sideslip Angle (Beta)
0.95	5.5 <b>d</b> eg	-4.1 deg
0.87	1.4 deg	-4.1 deg
0.36	, 14.6 deg	-14.5 deg
0.76	3.0 deg	0.0 deg

A negative sideslip angle was used as a direct result of the beta of most concern to the 4950TW. For the clean configuration, the direction the aircraft is sideslipped is not critical due to the symmetry of the aircraft, but for the dirty configuration this direction is crucial.

In the MERCURY flow solver, a negative beta implies that the nose of the aircraft yaws to the right (see Figure 3). Therefore, the decision to use negative beta conditions places the plate on the leeward side of the aircraft in yawed cases. This decision was made due to loading and aerodynamic concerns which were expected to be more significant than those for positive beta conditions. By changing signs of the results for the clean cases, it is possible to use the values for direct comparison with the dirty cases.

In addition to the change of beta, a change in Mach number from 0.95 to 0.87 was made for the dirty cases and led to an additional run of a clean case for direct comparison. This decision was made to add a flight condition clearly within the envelope as compared to those on the margin. With the plate and pylon mounted on the aircraft, it is unlikely that the aircraft will be flown at the outer limits of its Mach envelope. This change was made with the concurrence of Capt Wissler.

#### 6. CFD Results

Results from this analysis are presented in a variety of forms in the following sections.

# 6.1 Tables of Coefficients

Coefficients of Lift (CL), Drag (CD), Yaw (CY), as well as Moment Coefficients for Pitch (CMpitch), Yaw (CMyaw), and Roll (CMroll) were created for the clean and dirty solutions (Tables 4 and 5). In addition, incremental coefficient changes (Table 6) were calculated for the three cases where a direct comparison of flow conditions was possible.

The tabulated results show that the presence of the plate/pylon reduces the CL for moderate angles of attack. Although the pylon produces some lift, its influence on the right wing creates a net loss. In turn, the reduced lift on the right wing of the aircraft and the increased lift on the pylon appears to add to the nose up pitching moment and increases the negative rolling moment. In addition, the changed flow on the right wing leads to an increased positive yawing moment. By changing the downwash from the right wing, the airflow on the right side of the vertical tail is accelerated and produces additional yaw.

Results from the Mach 0.36 case illustrate a more complex flow field, and should be viewed in a qualitative manner. The severe angles of attack and sideslip produce massive separation on the left wing and the right side of the vertical tail and produces a wake off of the plate which follows the trailing edge of the wing (see Figure 11). The tabulated data shows a small increase in CL while CD increases significantly. It is important to remember that optical data will not be taken at this flight condition. Therefore, interest in this condition is purely for control and structural purposes.

Table 4: Coefficients for the Clean C-135

MACH	ALPHA	BETA	CL	CD	CY	${\tt CMpitch}$	$\mathtt{CMyaw}$	CMroll	AXIS
	(deg)	(deg)							
0.95	5.5	-4.1	0.4723	0.09357	0.02082	-0.25269	0.008940	0.020049	W
			0.4723	0.09184	0.02746	-0.25348	0.008940	0.001931	L S
			0.4790	0.04615	0.02746	-0.25348	0.008714	0.002779	В
0.95	1.4	-4.1	0.2831	0.04953	0.02575	-0.17381	0.009773	-0.016108	3 W
			0.2831	0.04757	0.02922	-0.17451	0.009773	-0.003639	S
			0.2841	0.04064	0.02922	-0.17451	0.009681	-0.003877	В
0.87	1.4	-4.1	0.3168	0.03357	0.03083	-0.10977	0.011409	-0.016773	W
			0.3168	0.03128	0.03315	-0.11069	0.011409	-0.008882	: S
			0.3175	0.02353	0.03315	-0.11069	0.011189	-0.009158	В
0.36	14.6	-14.5	0.5239	0.12461	0.01152	-0.20866	0.010332	0.071486	W
			0.5239	0.11776	0.04236	-0.21991	0.010332	0.016965	S
			0.5366	-0.01810	0.04236	-0.21991	0.005722	0.019022	В
0.76	3.0	0.0	0.3486	0.02182	0.00000	-0.10001	0.000000	0.000000	W
			0.3486	0.02182	0.00000	-0.10001	0.000000	0.000000	S
			0.3492	0.00355	0.00000	-0.10001	0.000000	0.000000	В

(Axis Systems: W = Wind, S = Stability, B = Body)

NOTE: For this analysis, a negative beta indicates the nose sideslips toward the right wing.

Table 5: Coefficients for the C-135 with Splitter Plate and Pylon

MACH	ALPHA	BETA	CL	CD	CY	${\tt CMpitch}$	$\mathtt{CMyaw}$	CMroll	AXIS
	(deg)	(deg)							
0.87	5.5	-4.1	0.4856	0.07605	0.03039	-0.14908	0.014029	-0.020329	e W
			0.4856	0.07368	0.03575	-0.15015	0.014029	-0.009618	3 S
			0.4904	0.02680	0.03575	-0.15015	0.013042	-0.010919	ЭВ
0.87	1.4	-4.1	0.3071	0.03356	0.03214	-0.10517	0.012059	-0.016904	1 W
			0.3071	0.03118	0.03446	-0.10611	0.012059	-0.009343	l S
			0.3078	0.02367	0.03446	-0.10611	0.011827	-0.009633	3 B
0.36	14.6	-14.5	0.5246	0.13415	0.04090	-0.21835	0.024171	-0.072048	B W
			0.5246	0.11964	0.07318	-0.22944	0.024171	-0.015082	2 S
			0.5378	-0.01647	0.07318	-0.22944	0.019588	-0.020688	3 B
0.76	3.0	0.0	0.3377	0.02318	0.00207	-0.08511	0.002170	-0.001423	3 W
			0.3377	0.02318	0.00207	-0.08511	0.002170	-0.001423	3 S
			0.3384	0.00548	0.00207	-0.08511	0.002093	-0.001534	4 B

(Axis Systems: W = Wind, S = Stability, B = Body)

NOTE: For this analysis, a negative beta indicates the nose sideslips toward the right wing. The plate is on the leeward side of the aircraft for all dirty cases.

Table 6: Incremental Coefficient Changes

# INCREMENTAL COEFFICIENT CHANGES BETWEEN CLEAN AND DIRTY CONFIGURATIONS DIRTY RESULTS MINUS CLEAN RESULTS (WIND-AXIS ONLY)

			(delta)	(delta)	(delta)	(delta)	(delta)	(delta)
MACH	ALPHA	BETA	CL	CD	CY	$\mathtt{CMpitch}$	$\mathtt{CMyaw}$	CMroll
	(deg)	(deg)						
0.87	1.4	-4.1	-0.0097	-0.00001	0.00131	0.00460	0.000650	-0.000131
0.36	14.6	-14.5	0.0007	0.00954	0.02938	-0.00969	0.013839	-0.000562
0.76	3.0	0.0	-0.0109	0.00136	0.00207	0.01490	0.002170	-0.001423

## 6.2 Particle Trace Plots for Clean Aircraft

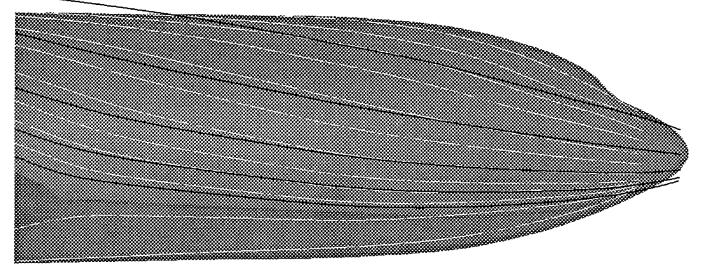
The particle trace plots for the clean cases are shown in Figures 4 - 8. The plots are provided to illustrate the angle of the air flow for a cross-section of the cases run.

The particle traces on the clean configuration shown in the following figures are plotted on and off the fuselage surface. The white streamlines are on the surface and the black streamlines are started approximately 1 foot off the surface (roughly the height of the pylon).

The plots show that the flow angularity is greater on the surface than off the surface in all cases. In general, the flow angle is approximately equal to the angle of attack (alpha) in the region where the plate will be attached, but it increases significantly as the flow approaches the wing.

These plots are only a sampling of the plots that were viewed by the investigators. The additional plots viewed were consistent with the trends demonstrated in those selected.

# Windward Side



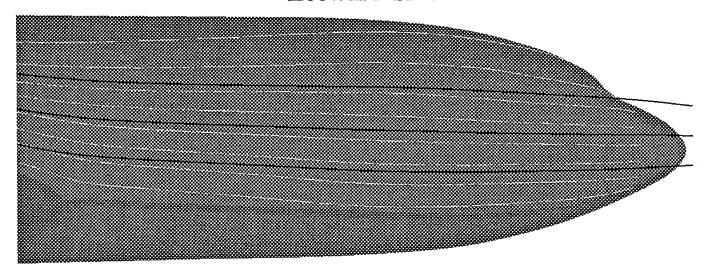
White Streamlines: on surface Black Streamlines: above surface

Mach # = 0.95 Alpha = 5.1 deg Beta = 4.1 deg

Figure 4: C-135 with Velocity Vectors on Surface (Windward Side)

(Mach=0.95, Alpha=5.5 deg, Beta=4.1 deg)

# Leeward Side



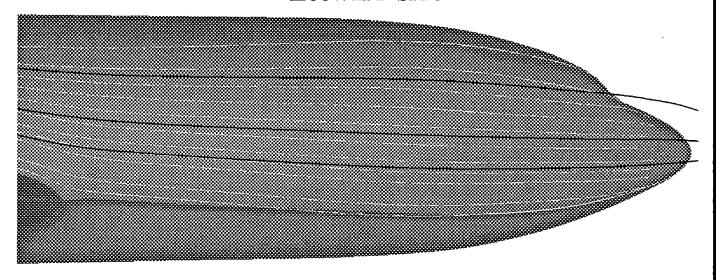
White Streamlines: on surface Black Streamlines: above surface

Mach # = 0.95Alpha = 1.4 deg Beta = -4.1 deg

Figure 5: C-135 with Velocity Vectors on Surface (Leeward Side)

(Mach=0.95, Alpha=1.4 deg, Beta=-4.1 deg)

# Leeward Side



White Streamlines: on surface Black Streamlines: above surface

Mach # = 0.87Alpha = 1.4 deg Beta = -4.1 deg

Figure 6: C-135 with Velocity Vectors on Surface (Leeward Side) (Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg)

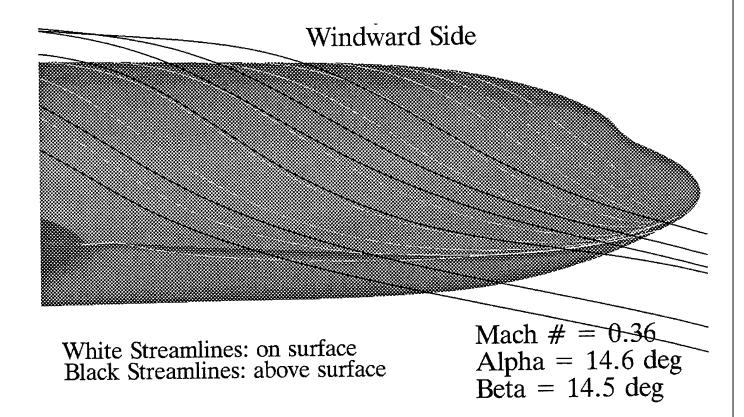
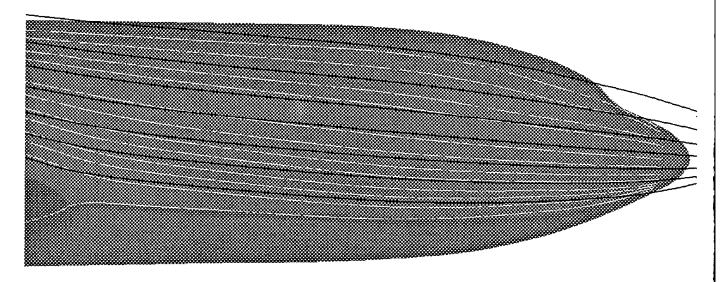


Figure 7: C-135 with Velocity Vectors on Surface (Leeward Side)

(Mach=0.36, Alpha=14.6 deg, Beta=14.5 deg)



White Streamlines: on surface Black Streamlines: off surface

 $\begin{array}{ll} \text{Mach} &= 0.76 \\ \text{Alpha} &= 3.0 \text{ deg} \\ \text{Beta} &= 0.0 \text{ deg} \end{array}$ 

Figure 8: C-135 with Velocity Vectors on Surface (Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

## **6.3** Particle Trace Plots for Dirty Aircraft

The Particle trace plots shown over the dirty aircraft in Figures 9 - 12 illustrate how the plate/pylon affects the flow over the aircraft. These plots should be used in conjunction with the pictures described in Section 6.4 to gain a better understanding of the air flow.

The particle traces shown in the following figures are plotted on the dirty aircraft in the region around the plate/pylon. The particles were released just upstream of the plate/pylon at a variety of heights and locations in an attempt to view the flow disturbance produced by the modification. Figures 9,10 and 12 show that the dirty flow produced by the plate/pylon swirls a little as it goes around the pylon and then passes under the horizontal stabilizer due to the downwash produced by the wing.

Figure 11 shows the particle traces which occur at severe angles of attack and sideslip. It appears that the flow wraps around the nose and fuselage without severe separation occuring on the plate. A vortical wake is produced off the trailing edge of the plate which passes outboard of the horizontal tail. Part of this wake also follows along the trailing edge of the wing. The lack of separated flow on the plate was initially surprising due to the amount of separated flow which appears on the leeward side of the vertical tail (see Figure 20). Two possible reasons for this lack of separation are the flow angle in the plate's proximity is significantly straightened by the presence of the fuselage, and/or the Euler formulation of the code did not capture the viscous effects which may cause separation.

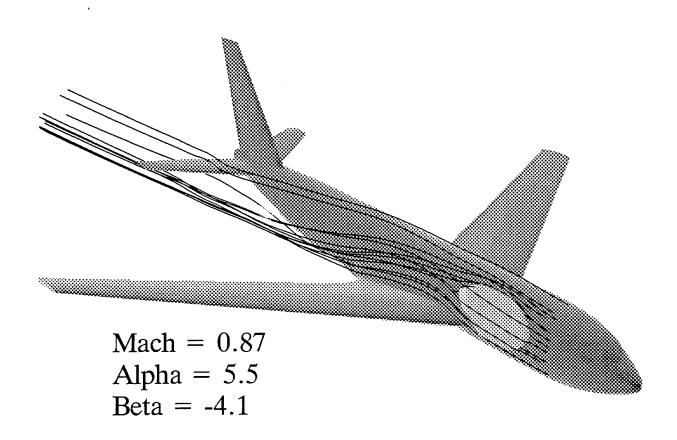


Figure 9: C-135 with Splitter Plate (Particle Traces)

(Mach=0.87, Alpha=5.5 deg, Beta=-4.1 deg)

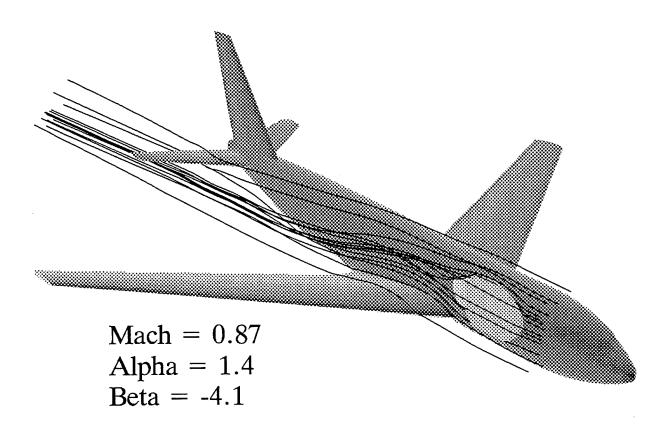


Figure 10: C-135 with Splitter Plate (Particle Traces)

(Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg)

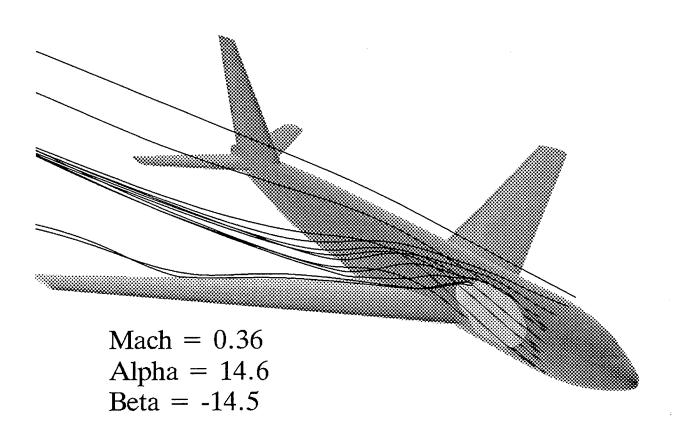


Figure 11: C-135 with Splitter Plate (Particle Traces)

(Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg)

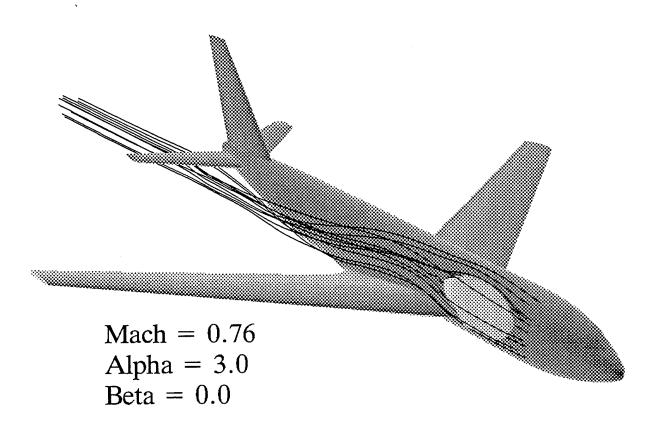


Figure 12: C-135 with Splitter Plate (Particle Traces)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

### 6.4 Mach Number Contour Plot on the Dirty Aircraft

The Mach number contour plots of the dirty configuration (Figures 13 - 23) illustrate the effects of mounting the plate/pylon. Note: the Mach number range differs on each plot to maximize the flow differentiation for each case.

The following plots contain two views of each dirty configuration run, and one view of the clean configuration where the flight conditions match. These plots should be used in conjunction with the previous particle trace plots to get a full picture of the flowfield. When viewing the contour plots the reader is reminded that for inviscid solutions of this type, peak Mach numbers tend to be higher with shocks occurring further aft.

#### 6.4.1 Mach=0.87, Alpha=5.5 deg, Beta=-4.1 deg

The first pair of plots, Figures 13 and 14, show the Mach=0.87, Alpha=5.5 degrees, Beta=-4.1 degrees case. On these plots, one sees uneven shocks on the wings and the accelerated flow on the leeward side of the vertical tail. Both of these conditions are attributed to the presence of sideslip. Also visible is the acceleration that occurs on the plate where the front begins to droop and on the top of the elliptical pylon. The stagnation that occurs on the pylon just aft of the plate indicates a very thin layer of stagnant flow that is present due to the blunt trailing edge of the plate. The wing aft of the plate/pylon shows lower Mach numbers near the root which leads to the reduced lift and increased rolling moment that the aircraft experiences at these flight conditions.

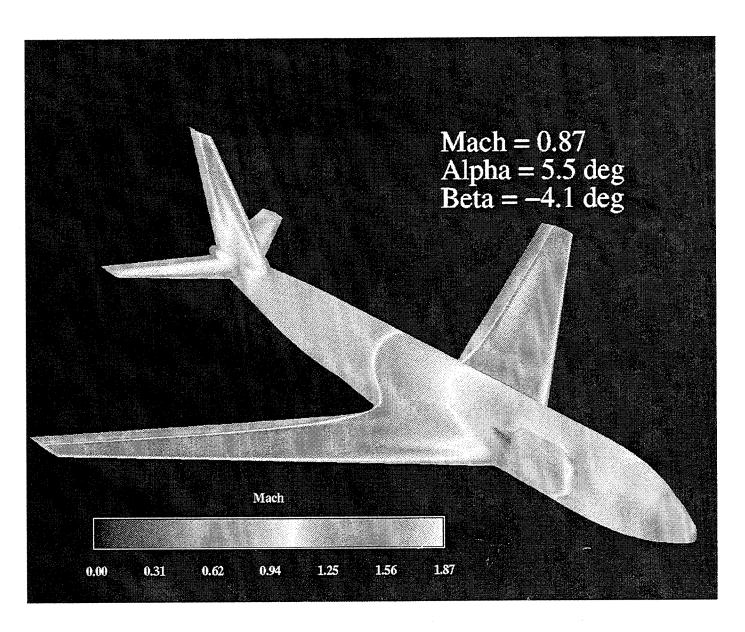


Figure 13: Dirty C-135 (Mach Contours)

(Mach=0.87, Alpha=5.5 deg, Beta=-4.1 deg)

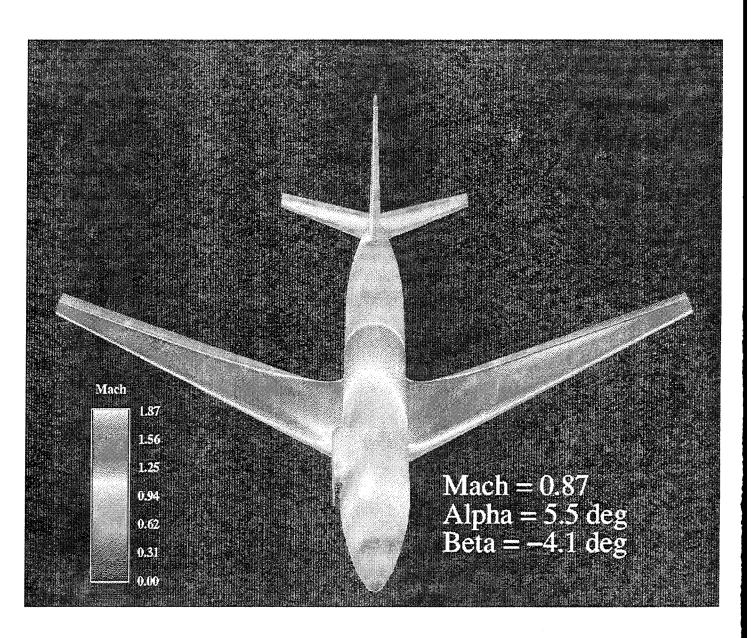


Figure 14: Dirty C-135 (Mach Contours, top view)

(Mach=0.87, Alpha=5.5 deg, Beta=-4.1 deg)

### 6.4.2 Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg

The next set of three plots, Figures 15-17, show the clean and dirty Mach=0.87, Alpha=1.4 degrees, Beta=-4.1 degrees cases. Comparison of Figures 15 and 16 illustrates the differences between the clean and dirty cases from a view above and forward of the aircraft.

The windward side of the aircraft and regions aft of the trailing edges experience negligible differences between the clean and dirty cases. The effect of the pylon in the dirty case includes increased Mach numbers on the top of the fuselage and reduced Mach numbers at the root of the leeward wing.

Figure 17 shows the side view where the top of the plate is visible. This view shows that the flow over the optical window region has a uniform Mach number. Additionally a weak shock is seen where the plate droops and a stagnation region is located aft of the plate's trailing edge.

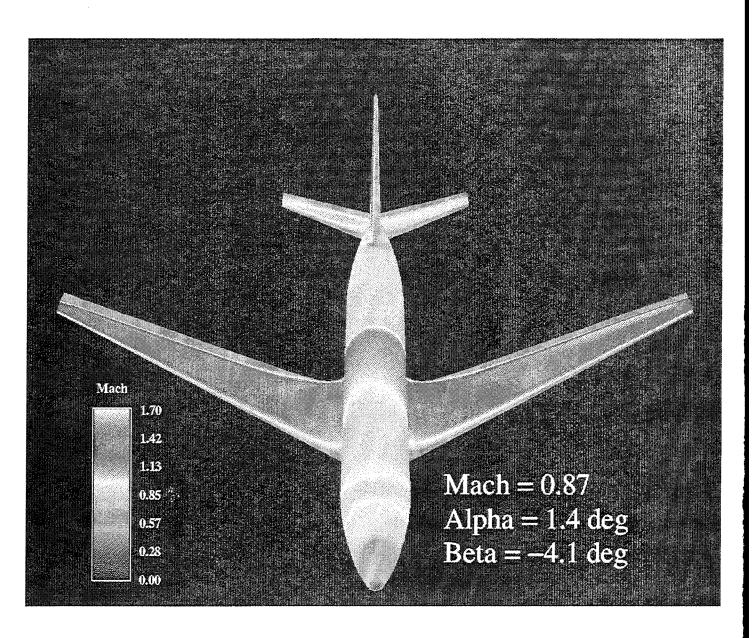


Figure 15: Clean C-135 (Mach Contours, top view)

(Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg)

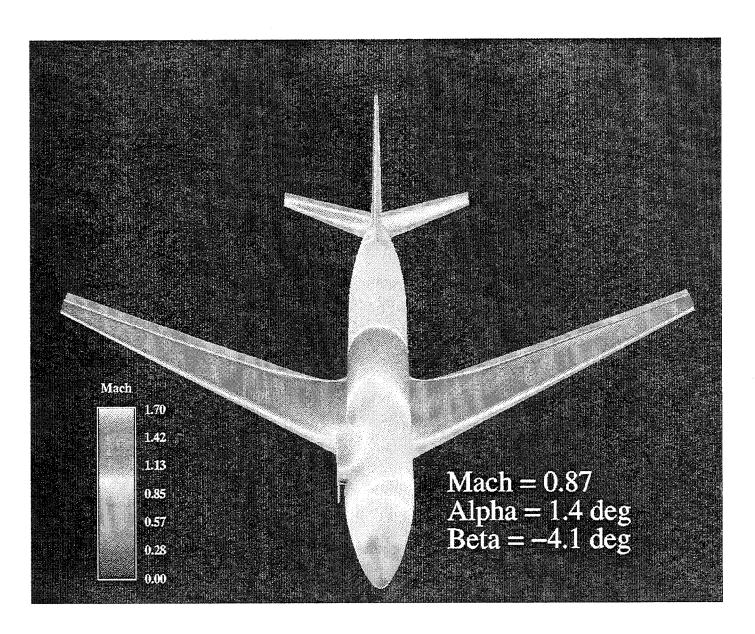


Figure 16: Dirty C-135 (Mach Contours, top view)

(Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg)

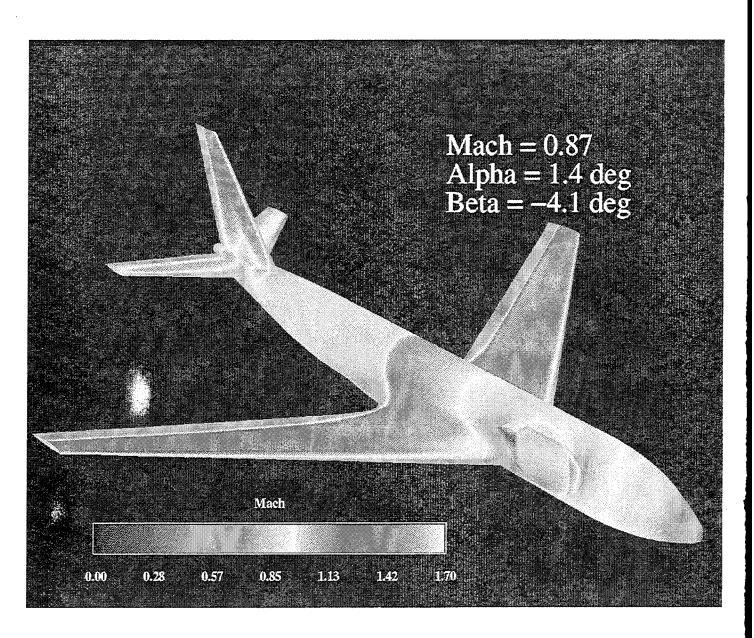


Figure 17: Dirty C-135 (Mach Contours)

(Mach=0.87, Alpha=1.4 deg, Beta=-4.1 deg)

#### 6.4.3 Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg

In the next set of three plots, Figures 18-20, show the clean and dirty Mach=0.36, Alpha=14.6 degrees, Beta=-14.5 degrees cases. Comparisons of Figures 18 and 19 illustrate the differences between the clean and dirty cases from a view above and forward of the aircraft.

The windward side of the aircraft and regions aft of the trailing edges experience negligible differences between the clean and dirty cases. The effect of the pylon in the dirty case is somewhat localized in the fuselage region but extends out the leeward wing, as previously seen with the particle traces.

Figure 20 shows the side view where the top of the plate is visible. This view shows that the flow over the optical window region has a slightly nonuniform Mach number distribution. One must remember when viewing these data that optical measurements will not be taken during this severe flight condition.

Results from this flight condition should be viewed with a critical eye due to the severity of the flight conditions and the amount of separated flow that is produced.

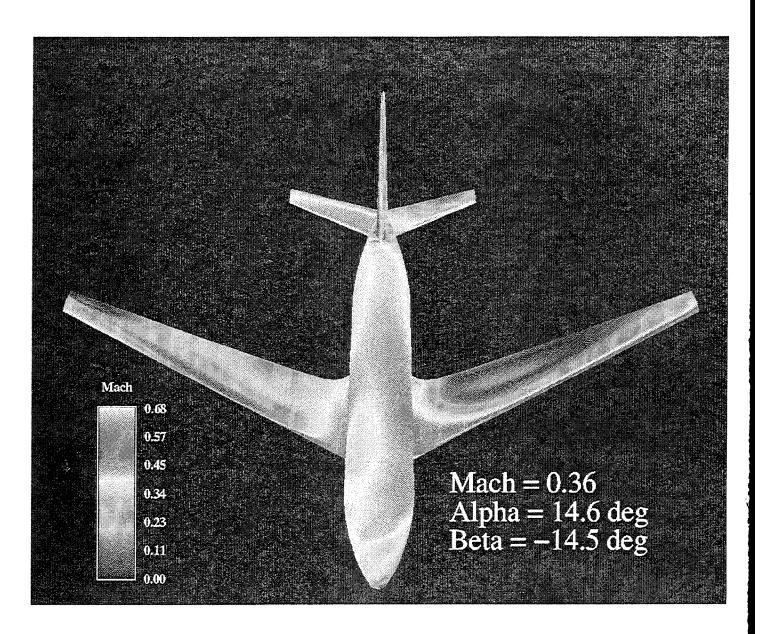


Figure 18: Clean C-135 (Mach Contours, top view)

(Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg)

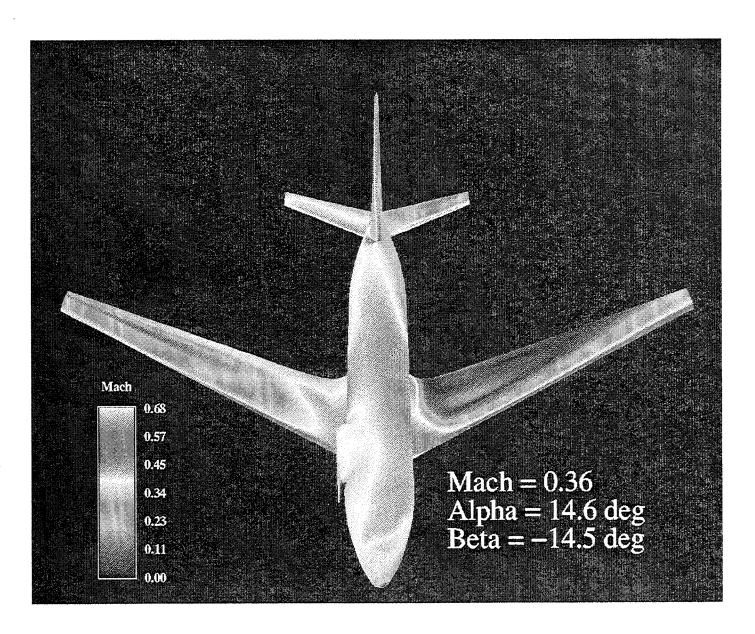


Figure 19: Dirty C-135 (Mach Contours, top view)

(Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg)

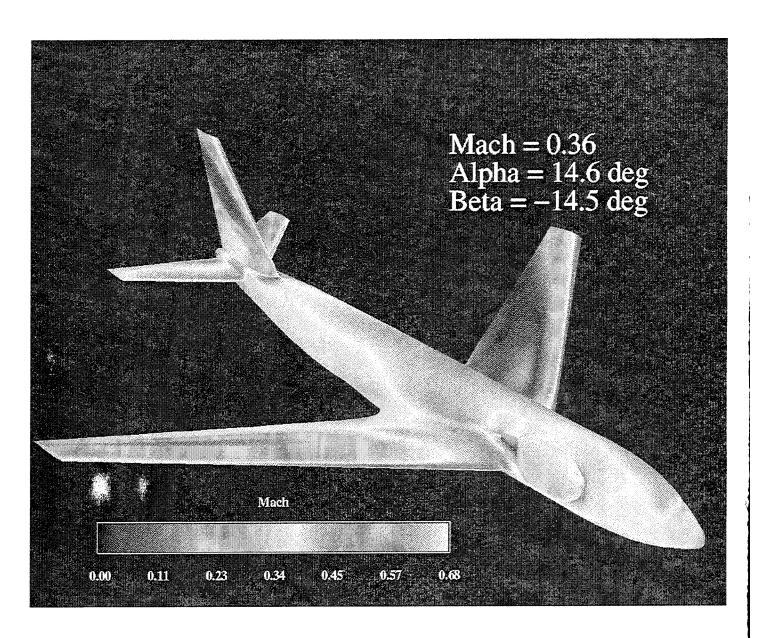


Figure 20: Dirty C-135 (Mach Contours)

(Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg)

### 6.4.4 Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg

In the next set of three plots, Figures 21-23, show the clean and dirty Mach=0.76, Alpha=3.0 degrees, Beta=0.0 degrees cases. Comparisons Figures 21 and 22 illustrate the differences between the clean and dirty cases from a view above and forward of the aircraft.

The windward side of the aircraft and regions aft of the trailing edges experience negligible differences between the clean and dirty cases. The effect of the pylon in the dirty case includes, increased Mach numbers on the top of the fuselage, reduced Mach numbers at the root of the wing (pylon side), and localized stagnations at the front and aft of the pylon.

Figure 23 shows the side view where the top of the plate is visible. Like previous plots, this view shows that the flow over the optical window region has a uniform Mach number distribution, the flow over the plate does not reach Mach 1, and that a stagnation aft of the plate is produced. (Figure 2 shows the position of the optical window in the plate.)

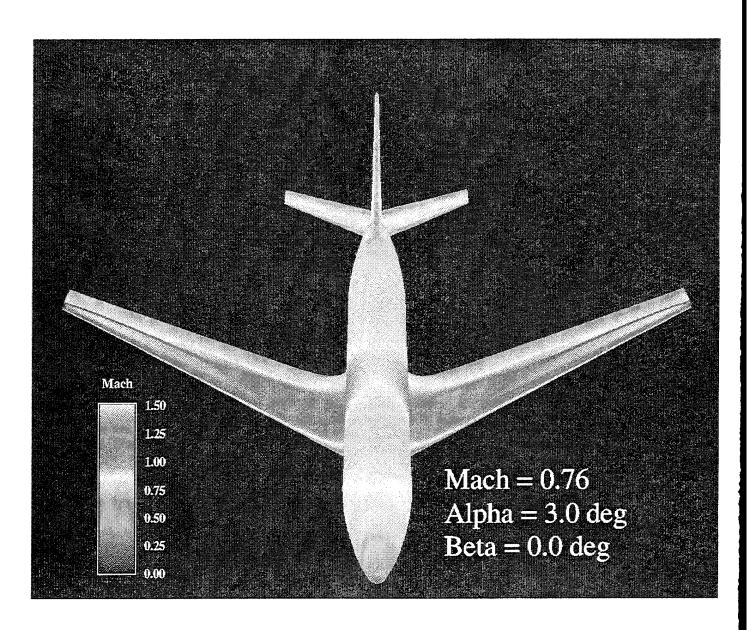


Figure 21: Clean C-135 (Mach Contours, top view)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

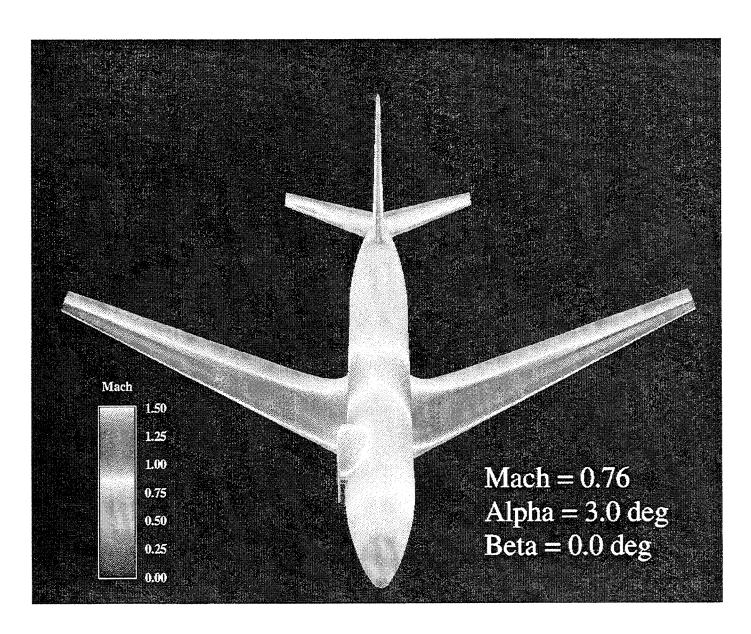


Figure 22: Dirty C-135 (Mach Contours, top view)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

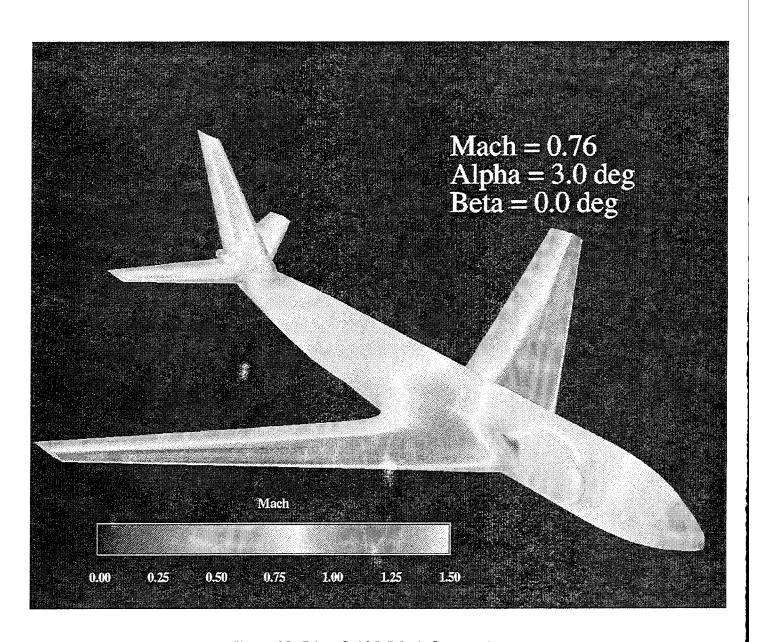


Figure 23: Dirty C-135 (Mach Contours)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

### 6.5 Pressure Coefficient Contour Plots on the Plate/Pylon

The pressure coefficient (Cp) contour plots on the plate/pylon shown in Figures 25 - 27 illustrate the aerodynamic loading. The plot for one case (Mach=0.87, Alpha=1.4, Beta==-4.1) was omitted due to its similarity to the other Mach 0.87 case. Small differences do occur, but the tabulated Cp values offer a better comparison tool than the visual plot. Note: the Cp range differs on each plot to maximize the flow differentiation for each case.

For each of the three Cp plots, the plate/pylon assembly is shown from four different view points without the aircraft. Figure 24 is included below to clarify the orientation of each view.

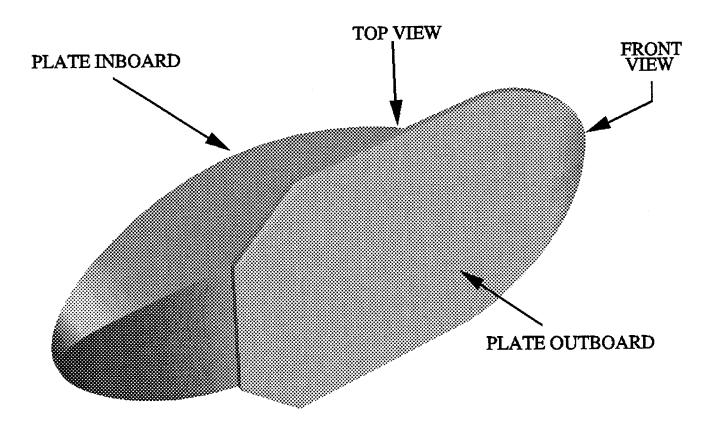


Figure 24: Splitter Plate and Pylon View Orientation

Figure 25 shows the splitter plate and pylon configuration for the Mach=0.87, Alpha=5.5 degrees, Beta=-4.1 degrees case. Flow stagnations are seen to occur on the leading edge of the plate and at the front and back of the elliptical pylon. Flow accelerations occur along the plate droop region and on the top and bottom of the pylon. These accelerated regions, indicated by the lower (more negative) Cps, are low pressure regions producing lift. A side force is produced by the combined low pressure on the outboard side of the plate and the high pressure on the inboard side. Due to the droop of the plate leading edge, the forces acting on the plate provide loads in the positive y and negative x axis directions with respect to the geometric coordinate system. As expected with an elliptical wing at angle of attack, the pressure distribution on the pylon indicates that lift is produced.

Figure 26 shows the Mach=0.36, Alpha=14.6 degrees, Beta=-14.5 degrees case. With the severity of the flight conditions, an unusual stagnation region is produced on the inboard side of the plate creating a fairly strong side force in the positive y-direction. Surprisingly, the flow on the outboard side of the plate does not appear to be separated and the distribution is uniform in the region of interest. Like the previous case the pylon acts as a lifting body at these flow conditions. It is expected that some separation would occur at this flight condition but the inviscid flow solver was unable to predict it.

Figure 27 shows the Mach=0.76, Alpha=3.0 degrees, Beta=0.0 degrees case. The Cp distribution shown in Figure 26 is very similar to the distribution shown in Figure 24. The Cp ranges of the two cases are different, however, the previous discussion holds for this case as well.

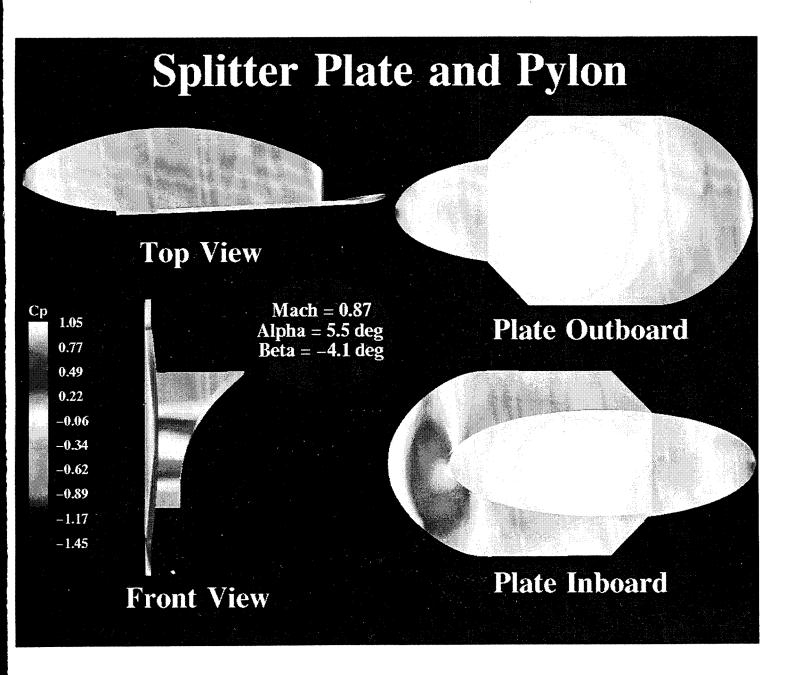


Figure 25: Splitter Plate and Pylon (Cp Contours)

(Mach=0.87, Alpha=5.5 deg, Beta=-4.1 deg)

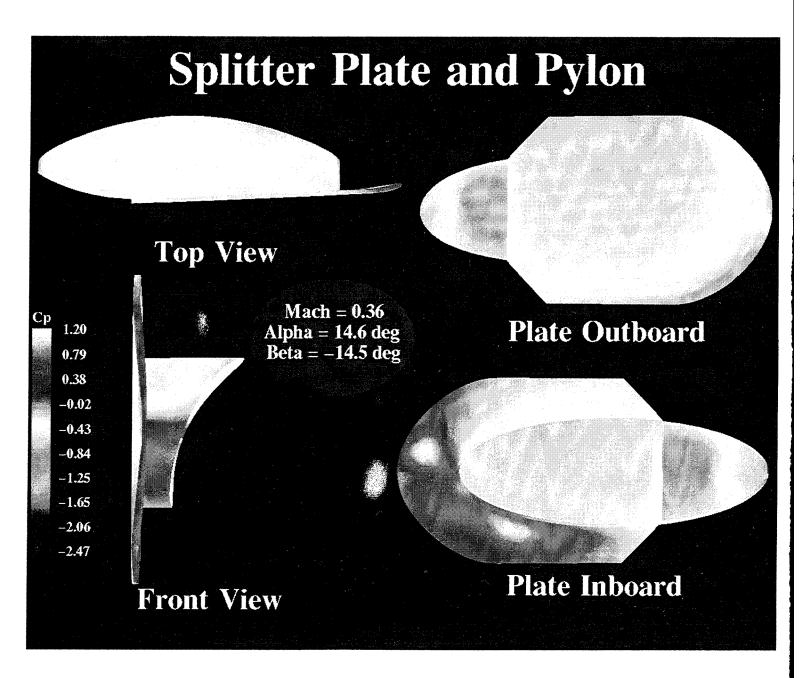


Figure 26: Splitter Plate and Pylon (Cp Contours)

(Mach=0.36, Alpha=14.6 deg, Beta=-14.5 deg)

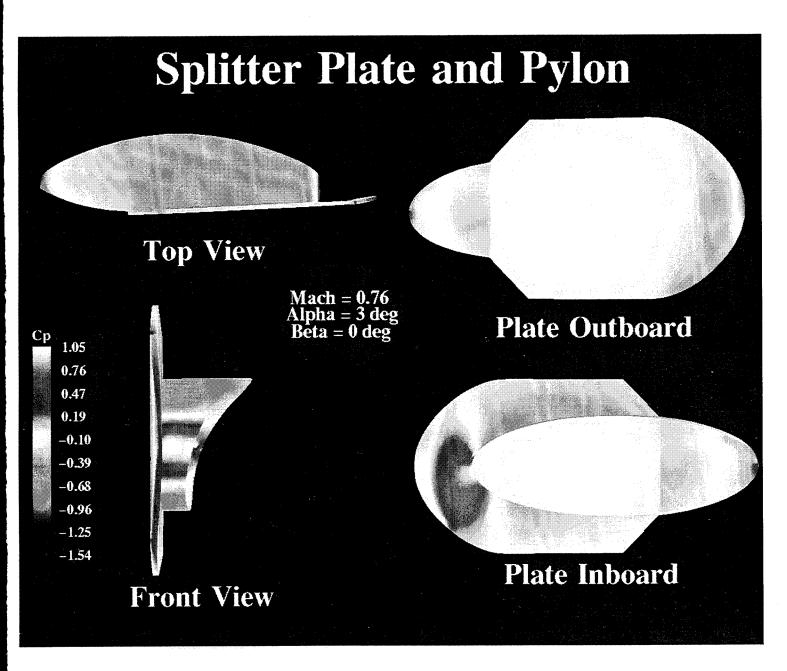


Figure 27: Splitter Plate and Pylon (Cp Contours)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

#### **6.6 PLOT3D Solution Files**

Solution files for all nine cases are available in PLOT3D format. These files can be used to extract additional information of interest. A picture only provides a snapshot of the results, while interactive viewing of the data provides a more complete insight into the results. (Interested parties should contact WL/FIMC for access to the solution files.)

### 6.7 Tabulated Cp Data on the Plate/Pylon

Tabulated Cp data on the plate/pylon for the four dirty cases is provided in the Appendix. This data was generated for conversion into loads data by the 4950TW. The table provides x, y, and z locations for each discrete panel center on the plate/pylon surfaces, the area of each panel, the unit normal components of each panel and the Cp at the centroid of each panel for the four dirty cases.

### **6.8** Cp Contour Plots on Symmetry Plane

Figures 28-29 illustrate two Cp contour plots on the symmetry plane of the clean aircraft. This data was generated for comparison with published data on the C-135 aircraft [5].

Figure 28 shows a side view of the flow at the aircraft symmetry plane for the Mach=0.76, Alpha=3.0 degree case. Visible are the stagnations at the nose, the canopy and the leading edge of the vertical tail. Also visible are the low pressure regions produced by the flow accelerating over the canopy and wing.

Figure 29 is a close-up of the nose region of the same case. The Cp range has changed to accentuate the flow features present. This plot was generated to answer some questions about positioning probes.

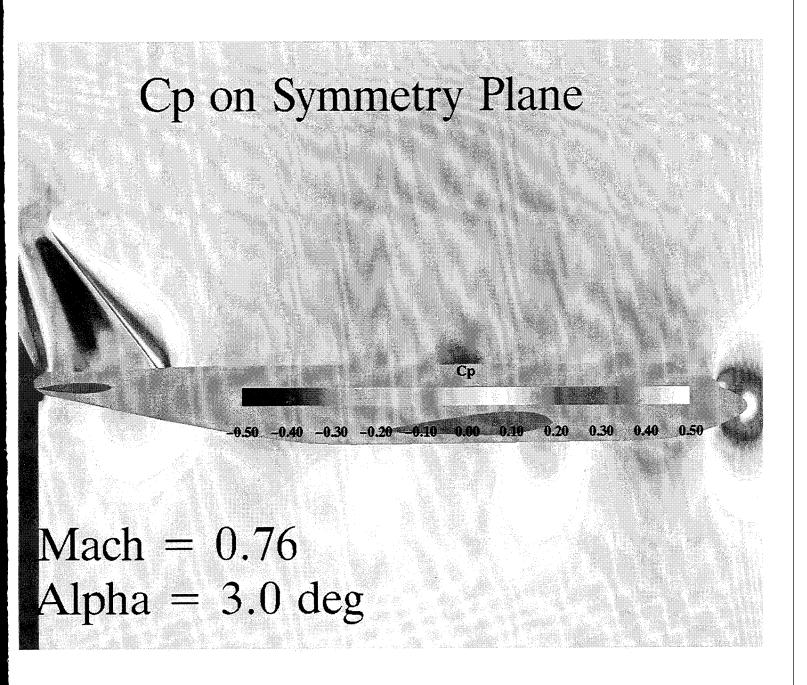


Figure 28: Cp on Symmetry Plane (Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

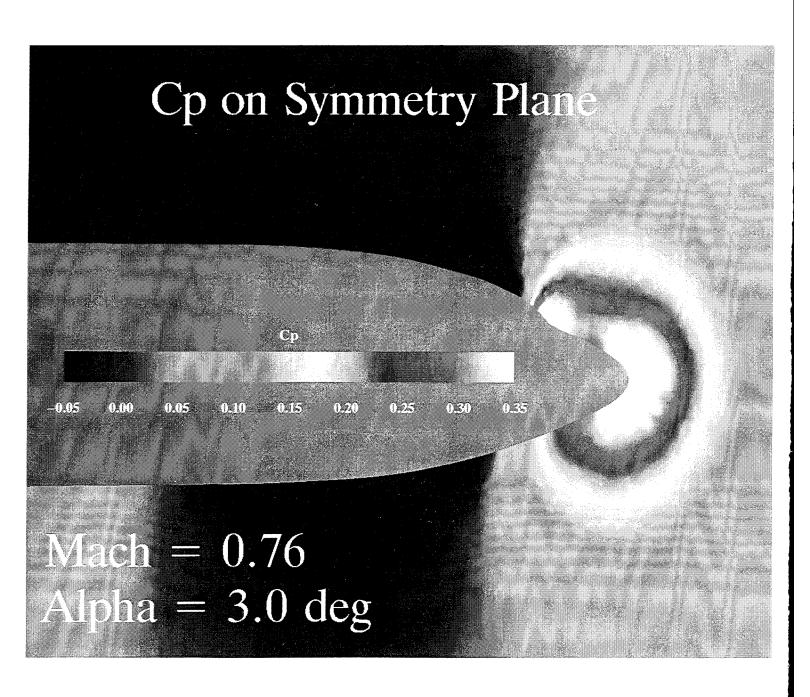


Figure 29: Cp on Symmetry Plane (Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

## 6.9 Fuselage X-station Plots

Plots of Cp vs. X-station and Mach number vs. X-station are provided along the clean configuration centerline in Figures 30 - 33. These plots provide data on both the top and bottom of the fuselage for the reader's convenience.

# Cp vs. X-station (fuselage top centerline)

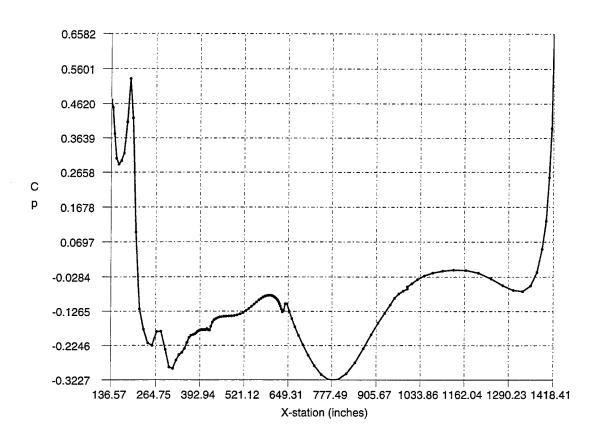


Figure 30: Cp vs. X-station (fuselage top centerline)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

# **Cp vs. X-station** (fuselage bottom centerline)

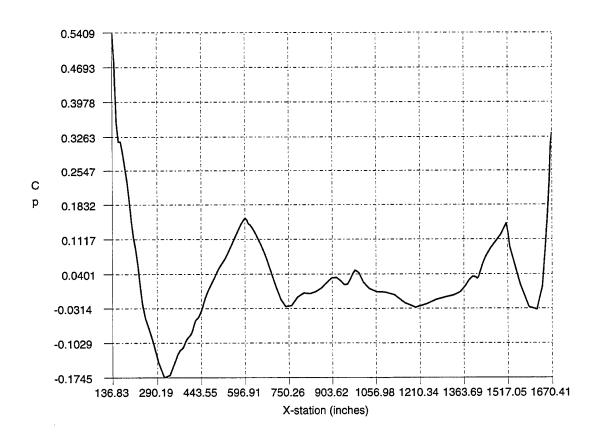


Figure 31: Cp vs. X-station (fuselage bottom centerline) (Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

### Mach Number vs. X-station (fuselage top centerline)

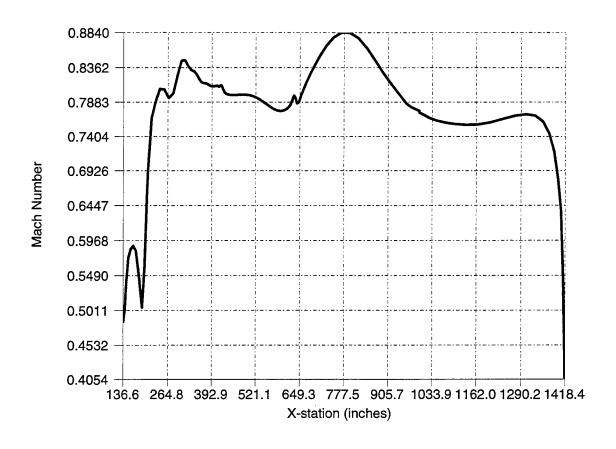


Figure 32: Mach Number vs. X-station (fuselage top centerline)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

# Mach Number vs. X-station (fuselage bottom centerline)

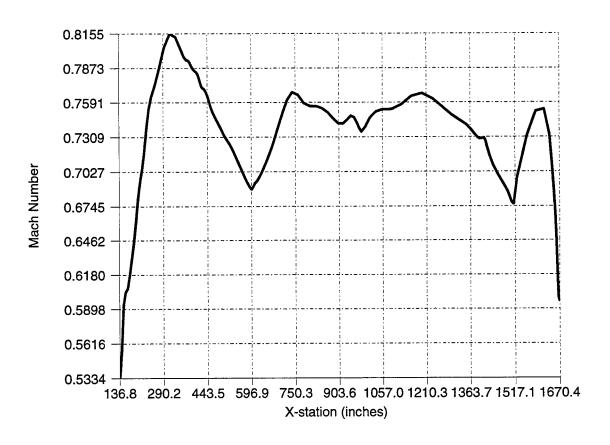


Figure 33: Mach Number vs. X-station (fuselage bottom centerline)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

### 6.10 Plate X-station Plots

Plots of Cp vs. X-station and Mach number vs. X-station are provided along the centerline of the plate for the Mach = 0.76 case. Figures 34 and 35 show wind tunnel results and are provided for comparison.

# **Cp vs. X-station (plate top centerline)**

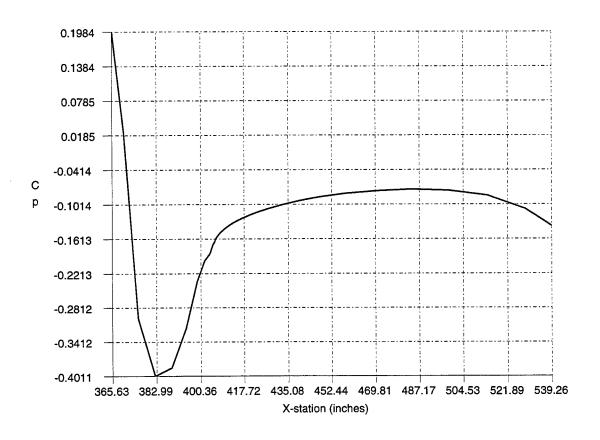


Figure 34: Cp vs. X-station (plate top centerline)

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

## Mach Number vs. X-station (plate top centerline)

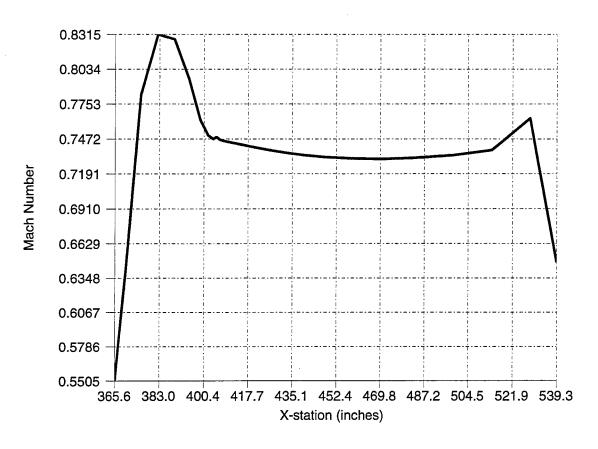


Figure 35: Mach Number vs. X-station (plate top centerline) (Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

### **6.11** Particle Trace Plot

The streamlines for the Mach = 0.76 case are plotted to illustrate the particle paths near the nose of the C-135. The position listed on the plot in Figure 36 is of interest to the ABL SPO due to an instrument probe location.

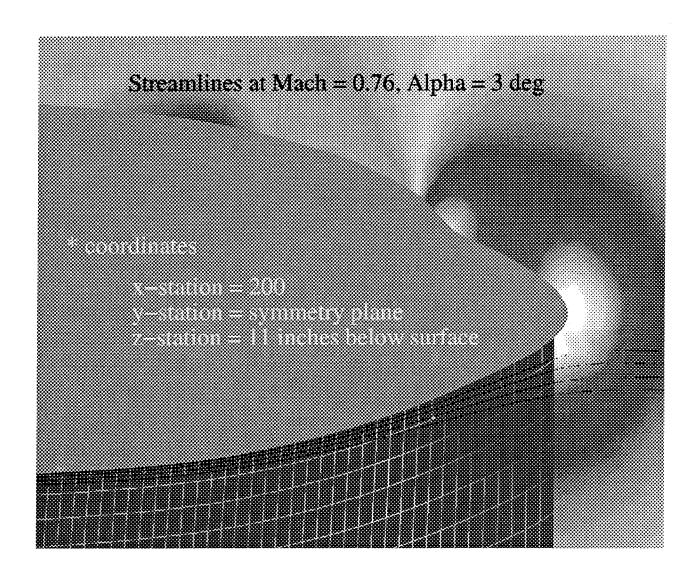


Figure 36: Streamlines under Aircraft Nose

(Mach=0.76, Alpha=3.0 deg, Beta=0.0 deg)

#### 7. CFD Results Summary

The analysis provided here is very general in nature and is intended to bring several observations to light. More detailed analysis would require additional direction by the ABL SPO to focus on the specific requirements of the analysis. In general, gaining an understanding of specific areas of concern is best accomplished through discussion and interactive viewing of the data.

From the previously presented results, it is obvious that the plate/pylon have a significant effect on the air flow over the aircraft. The tabulated results show that the presence of the plate/pylon reduces the CL for moderate angles of attack. Although the pylon produces some lift, its influence on the right wing creates a net loss. In turn, the reduced lift on the right wing of the aircraft and the increased lift on the pylon appears to add to the nose up pitching moment, increases the negative rolling moment, and increases the positive yawing moment. The Mach number plots provided show reduced Mach number over the right wing and also shows that the right horizontal stabilizer is experiencing increased lift perhaps due to reduced downwash of the right wing. The flow over the plate and pylon can also be seen in the Mach number plots. These computations show that the pylon has significant transonic and low supersonic flow over most of its surface for cases where the freestream Mach number exceeds 0.76 and may experience mild shocks, especially at the 5.5 degree angle of attack case. The plate sees a local acceleration on the leading edge droop and of the plate and a weak shock forms for the higher freestream Mach number cases. This shock increases in strength as the Mach number increases but the location remains stationary. The flow over the pylon aft of the plate resembles a recirculation region. This flow region occurs only near the surface and may be due to the modelling of the plate thickness. The aft edge of the plate was purposefully left with a rearward facing step to fix the location of separation at the back of the plate.

Results from the Mach 0.36 case illustrate a more complex flow field, and should be viewed in a qualitative manner. The severe angles of attack and sideslip produce massive separation on the left wing and the right side of the vertical tail. This in turn produces a wake off of the plate which follows the trailing edge of the wing (see particle trace plot). Surprisingly, the flow on the outboard side of the plate does not appear to be separated. From the particle trace it appears that

the flow wraps around the nose and fuselage without separating. It is important to remember that optical data will not be taken at this flight condition. Therefore, interest in this condition is purely for control and structural purposes.

The Cp plots on the plate should be used in conjunction with the tabulated Cp data. The cases plotted all show a uniform Cp distribution in the region where the optical window is located. The tabulated Cp data was provided to the 4950 Test Wing to calculate loading on the plate and pylon. Results of their work are included in Section 9 of this report and in Reference 3.

### 8. Tunnel Test Comparison

In April 1993, a wind tunnel test was conducted on four plate/pylon configurations in the WL/FIME Trisonic Gasdynamics Facility. This facility is a closed circuit, variable density, continuous flow wind tunnel with an operating Mach number range of 0.23 to 3.0.

For the tests conducted, a freestream Mach number of 0.7 was used. The 0.1 scale plate/pylon models were mounted on the side wall of the 2-foot test section and were able to pivot to change the angle of attack (AoA). No sideslip cases (yaw angle not equal to zero) were possible due to the test setup. The main purpose of the wind tunnel test was to measure the boundary layer thickness in the optical window region. In addition to collecting boundary layer data, static pressure ports collected data on the splitter plate centerline, and tufts and oil flows were used for flow visualization. (For complete information on the wind tunnel test, the reader is directed to Reference 1.)

For purposes of comparison with the CFD analysis, the baseline plate tested in the tunnel is a scaled version of the geometry used for the CFD analysis. One must remember that the tunnel model is mounted on a flat wall while the CFD configuration has a plate/pylon mounted on a curved fuselage. An inspection of the CFD and tunnel pressure data shows consistently uniform flow in the optical window region, and similar trends in the data. Direct comparison of the pressure values is not included due to the differences in flight conditions and geometry. Inclusion of the wind tunnel data in this report is solely for a qualitative comparison of streamline data with the calculated results.

Courtesy of Dr.James Van Kuren, the following two pictures (Figures 37 and 38) are included for comparison with the CFD results. Figure 37 shows tufts attached to the outboard surface of the plate. These tufts show the uniformity of the flow on the plate. Figure 38 shows an oil flow at an angle of attack of 2.5 degrees. Of particular interest in this picture is the stalled region (indicated by the stationary oil) on the pylon aft of the plate. This region is caused by the bluntness of the plate's trailing edge, and is consistent with findings in the CFD results (see Figure 39).

Although the flow conditions are slightly different for Figures 38 and 39, the angles of the oil

flow on the plate are remarkably similar. The oil flow on the pylon in Figure 39 indicates that the flow is also stalled in the computational solution as mentioned above.

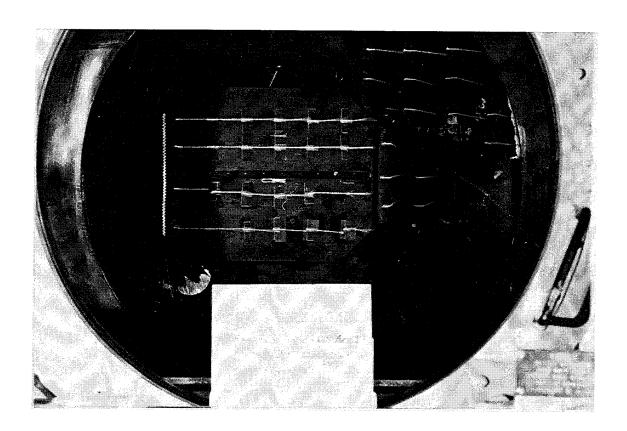


Figure 37: Tufted Splitter Plate in Tunnel

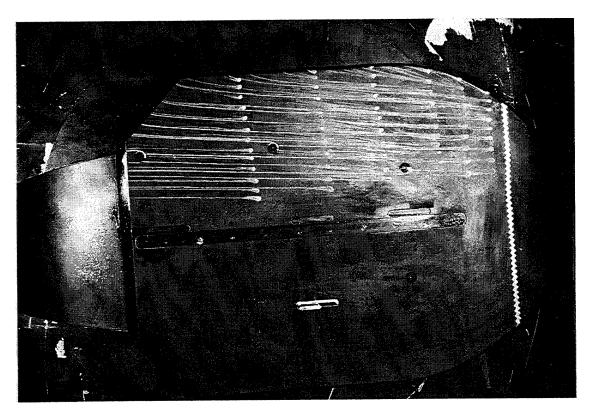


Figure 38: Splitter Plate Oil Flow in Tunnel (Mach=0.7, Alpha=2.5 deg)

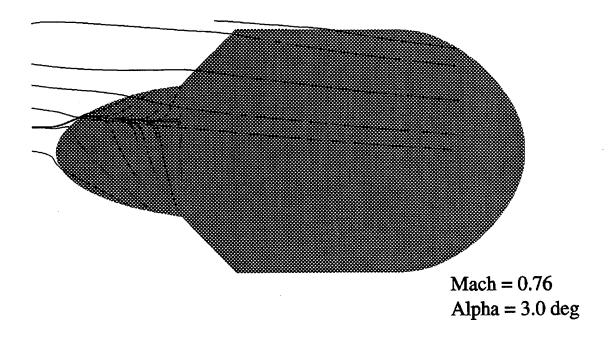


Figure 39: CFD Oil Flow on Splitter Plate

### 9. Plate and Pylon Loading

From the CFD analysis performed, pressure coefficient (Cp) data was extracted for use by the 4950 Test Wing. The Appendix contains the tabulated Cp data for the panels which make up the plate and pylon surfaces. This data was provided to the 4950 Test Wing where it was used to produce the loads and moments which act on the plate and pylon. From the Cp data, pressure loads (in lbs) were calculated on surface panels by using the definition of Cp, the area of the surface panel, and the unit normal. By summing the panels of interest, total loads and their directions were calculated. For the moment calculations, the loads were summarized around: fs 450, bl 84.5, and wl=256.

Courtesy of Mr Dave Bushroe of the 4950 Test Wing the following tables, seven thorugh eleven, have been included to complete the results.

Table 7: Test Condition Summary

Condition	Altitude	Vе	Mach	Beta	Alpha	q	q
	(feet)	(KEAS)		(deg)	(deg)	(psf)	(psi)
1	23800	360	0.87	-4.1	5.5	437.76	3.04
2	23800	360	0.87	-4.1	1.4	437.76	3.04
3	0	240	0.36	-14.5	14.6	191.98	1.33
4	45000	189	0.76	0.0	3.0	124.53	0.86

(KEAS = Knots Equivalent Airspeed, q = dynamic pressure)

In the following tables, eight through eleven, the first four surfaces represent the sides of the pylon, the next two represent the inboard plate surfaces, and the remaining surfaces represent the outboard plate surfaces and the top of the pylon.

Table 8: Pressures and Moments for Condition 1

Pressure and Moments on Plate and Pylon Surfaces
(Alt=23800 ft, Ve=360 keas, M=.87, Beta=-4.1 deg, Alpha=5.5 deg, q=3.04 psi)

Surface	Area	рx	ру	pz	mxx	myy	mzz
L A Py	751.0	10.6	0.6	-173.6	986.1	16122.5	-157.2
F Py B	2099.1	476.6	0.0	-918.3	4111.6	43926.9	4146.8
U A Py	2152.6	142.5	0.5	944.5	-18460.7	-85206.8	3061.0
F Py U	5923.6	-688.8	2.0	6469.4	-121879.8	-139361.0	-11769.4
Py Tot	10926.3	-59.1	3.1	6322.0	-135242.7	-164518.4	-4718.7
L Pl B	5231.6	-63.9	764.8	0.0	-21792.1	83.8	-160241.2
U Pl B	5224.2	48.1	-2402.5	0.0	129566.5	4002.9	-89673.2
Ру Т А	Pl 3182.6	89.1	1020.4	0.0	-120.5	-131.7	89095.0
U Pl T	5228.7	-172.0	3110.2	-0.2	-99874.1	-4651.7	-132871.9
M Pl T	7749.1	-63.4	1814.6	0.0	-3276.5	-112.4	23616.6
L Pl T	5236.0	-207.7	3665.2	0.7	124729.6	5370.8	-176362.0
Pl Tot	31852.3	-389.8	7972.8	0.6	119293.8	4561.5	-426436.5
Totals		-448.9	7975.9	6322.6	-15948.8	-159956.9	-431155.3

(All table values have been rounded to the nearest decimal place for tabulation. For surface names the following abbreviations are used: L=Lower, A=Aft, F=Forward, U=Upper, B=Bottom, T=Top, M=Middle, Py=Pylon, Pl=Plate)

Table 9: Pressures and Moments for Condition 2

Pressure and Moments on Plate and Pylon Surfaces
(Alt=23800 ft, Ve=360 keas, M=.87, Beta=-4.1 deg, Alpha=1.4 deg, q=3.04 psi)

Surface	e Area	рx	ру	pz	mxx	myy	mzz
L A Py	751.0	-39.5	0.4	-57.7	150.7	4144.9	-664.5
F Py B	2099.1	140.4	0.1	1682.7	8894.7	26964.5	1983.6
U A Py	2152.6	138.0	0.5	881.8	-17411.1	-78935.8	2965.6
F Py U	5923.6	-174.7	1.7	5899.9	-114582.7	-160239.7	-5412.2
Py Tot	10926.3	84.1	2.6	5041.3	-122948.4	-206045.5	-1127.5
L Pl B	5231.6	-47.1	-888.0	0.0	-86105.8	-1666.8	-116937.4
U Pl B	5224.2	-5.9	-1313.4	0.1	99029.9	2574.7	-130833.4
Py T A	Pl 3182.6	-54.6	161.8	0.0	-321.3	<del>-</del> 28.3	3783.7
U Pl T	5228.7	-230.8	4006.2	-0.5	-129342.5	-8122.9	-181044.8
M Pl T	7749.1	-69.9	2000.9	0.0	-4021.1	-140.5	19696.6
L Pl T	5236.0	-239.7	4002.6	1.1	116673.1	5762.2	-203232.4
Pl Tot	31852.3	-648.2	7970.1	0.7	-2087.9	176.3	-608565.7
Totals		-584.1	7972.7	5041.9	-125036.3	-205867.1	-609693.2

(All table values have been rounded to the nearest decimal place for tabulation. For surface names the following abbreviations are used: L=Lower, A=Aft, F=Forward, U=Upper, B=Bottom, T=Top, M=Middle, Py=Pylon, Pl=Plate)

Table 10: Pressures and Moments for Condition 3

Pressure and Moments on Plate and Pylon Surfaces
(Alt=0 ft, Ve=240 keas, M=.36, Beta=-14.5 deg, Alpha=14.6 deg, q=1.33 psi)

Surface	Area	px	ру	pz	mxx	myy	mzz
L A Py	751.0	141.3	1.0	-438.8	2868.8	43175.3	1238.4
F Py B	2099.1	269.4	-0.2	-168.5	367.6	29746.9	1967.7
U A Py	2152.6	320.4	0.7	1026.0	-19259.9	-100850.3	6002.8
F Py U	5923.6	-397.6	0.9	2991.6	-56019.3	-74356.1	-5357.8
Py Tot	10926.3	333.4	3.4	3410.3	-72042.7	-102284.1	3851.1
L Pl B	5231.6	-114.6	2398.1	0.0	77212.4	3272.0	-97166.6
U Pl B	5224.2	-53.0	-622.0	0.0	47528.9	524.1	-70578.4
РуТА	Pl 3182.6	501.4	2918.7	0.1	-3819.3	587.4	291952.4
U Pl T	5228.7	-101.5	1581.5	-0.1	-52008.6	-2576.5	-59949.4
M Pl T	7749.1	-75.0	2146.9	0.0	2237.3	78.0	47561.0
L Pl T	5236.0	-188.0	3392.0	-0.4	129446.1	5634.8	-95113.7
Pl Tot	31852.3	-30.7	11815.1	-0.3	200596.7	7519.8	16705.3
Totals		302.8	11817.5	3410.0	128554.0	-94764.3	20556.3

(All table values have been rounded to the nearest decimal place for tabulation. For surface names the following abbreviations are used: L=Lower, A=Aft, F=Forward, U=Upper, B=Bottom, T=Top, M=Middle, Py=Pylon, Pl=Plate)

Table 11: Pressures and Moments for Condition 4

Pressure and Moments on Plate and Pylon Surfaces

(Alt=45000 ft, Ve=189 keas, M=.76, Beta=0 deg, Alpha=3 deg, q=.865 psi)

Surface	Area	рx	ру	pz	mxx	myy	mzz
L A Py	751.0	9.3	0.3	-67.7	401.8	6416.9	9.6
F Py B	2099.1	107.6	0.0	-344.8	1667.3	12592.0	1004.9
U A Py	2152.6	72.7	0.2	372.4	-7079.2	-34332.9	1447.3
F Py U	5923.6	-222.0	0.7	2067.4	-39330.3	-46416.4	-3732.3
Py Tot	10926.3	-32.5	1.1	2027.3	-44340.5	-61740.4	-1270.5
L Pl B	5231.6	-6.8	-60.2	0.0	-14467.5	-376.0	-34105.1
U Pl B	5224.2	34.0	-956.7	0.0	45337.2	1640.6	-8762.0
Ру Т А Р	Pl 3182.6	53.1	424.4	0.0	39.6	-16.7	39477.6
U Pl T	5228.7	-36.7	732.4	0.0	-25268.1	-1086.0	-22039.4
M Pl T	7749.1	-21.8	625.0	0.0	-1415.3	-49.4	12333.8
L Pl T	5236.0	-45.8	780.5	0.0	24342.9	1173.9	-35501.8
Pl Tot	31852.3	-24.0	1545.5	0.0	28568.7	1286.4	-48596.8
Totals		-56.5	1546.6	2027.3	-15771.8	-60454.0	-49867.3

(All table values have been rounded to the nearest decimal place for tabulation. For surface names the following abbreviations are used:

L=Lower, A=Aft, F=Forward, U=Upper, B=Bottom, T=Top, M=Middle, Py=Pylon, Pl=Plate)

## 10. References

- [1] Van Kuren, James T. "ABL AACT Splitter Plate, 0.1 Scale Wind Tunnel Test," April 1993
- [2] Emsley H. T. "I3G/VIRGO, Interactive Graphics for Geometry Generation and Visual Interactive Rapid Grid Generation, User's Manual," WL-TM-91- 316.
- [3] Emsley H. T. "PLUTO 3-D Grid Generator, User's Manual," WL-TM-91-312.
- [4] Strang W. Z. "Mercury User's Manual," AFWAL-TM-88-217.
- [5] Boeing Company "External Internal Loads for the C-135 Airplane," D6-7267, April 1961.

# Appendix

(Cp data on plate and pylon surfaces)

## THIS TABULATED DATA PROVIDES THE PRESSURE COEFFICIENT DATA OF ALL PANELS ON THE PLATE AND PYLON SURFACES

: THE CENTROID COORDINATE LOCATION OF THE PANEL : THE SURFACE AREA OF THE PANEL EZ: THE UNIT NORMAL COMPONENTS

X,Y,Z: AREA: EX,EY,EZ:

PRESSURE COEFFICIENT AT (MACH = 0.76, ALPHA = 3.0, BETA = 0.0)
PRESSURE COEFFICIENT AT (MACH = 0.87, ALPHA = 5.5, BETA = -4.1)
PRESSURE COEFFICIENT AT (MACH = 0.36, ALPHA = 14.6, BETA = -14.5)
PRESSURE COEFFICIENT AT (MACH = 0.87, ALPHA = 1.4, BETA = -4.1) CP1 CP2 CP3 CP4

#### pylon aft lower

0886999221828691878889412988277182978809971889227782988894448882771555735691889286978899742442429221881678497888894112281601237889888941888937188988889418889371889888894188893718898888941888893718898888941888893718888894888889888889888888988888889888888	0727559326380429776640308212264810712929851210689773998412225793789914153350821226488107129298512106897776566666666777751443232110695778448333867776666688870998655787877777777777777777777777666666666	4669938692119275999982399889554887993727885748653657778676451938619918822119925222211939546663933711111999582339889554887993727855748653657778667658933771111199958233988937778557486536677778667657877878787878787878787878787	9111399915559344944703216488634425431777112411132467952333442385115048275225433177116421115050368693832198879421783276193585034421319355536693832193887133165476267638531988879421715050505050505050505050505050505050505	1890211387846194609135922405846617967781630520792595407837034409311100943 00268725788936694877837748041477057857116673746929127253449546778375259540792573574695291272534495737748959628559359 X733552957257899962295296182795725789962253449566778899962257455667788559359 X7335529571852957257456678899962959999912233445667788996285748559359 X73355296788899999999122334456667788999999912233445666778899999999999999999999999999999999	EY30303457 0.0033457 0.0033457 0.0033559 0.00335594 0.003305594 0.003275 0.0032	40150600406497592466493556810444718593656346413889717602205663446759476593686277665247862797111078757666247871110787575766211110787576662478711107875757662787766787767877678776787767877	03504410497643818124414886258824297438510499764380229648876429773438313812484448862587853799211133487494976853141049976438022336556360424	415161007 42464 4151610107 41516 41516 416	57602145548310028518888863460207155846020145584503076727758856465201455865667676776522	-0.00644 -0.00930 -0.02188 -0.01188 0.06916 0.13350
547.6918 559.9279 578.0586 584.38227 593.0198 598.140	80.6647 79.4623 76.7847 75.5828 74.5159 73.57764 72.0818		27.7558627 7558627 7558627 7558627 96.24538825 14.398531941 97.6852999 97.6852999 97.68529 97.68529 97.685	0.2872 0.3509 0.4173 0.4881 0.5581	0.0017 0.0022 0.0035 0.0055 0.0060 0.0006 -0.0106 -0.0055	-0.9745 -0.9579 -0.9579 -0.9587 -0.82982 -0.73491 -0.65340 -0.43030 -0.43030 -0.1637 -0.1137	-0.12504 -0.03368 -0.007140 0.097140 0.14774 0.23822 0.26322 0.30949 0.39577 0.46126	-0.09212 -0.09505 -0.109305 -0.109302 -0.0712802 -0.0712802 -0.0138250 0.138250 0.281993 0.385444 0.563361 0.56361 0.56361	-0.47570 -0.39202 -0.25197 -0.27551	-0.03782 -0.023782 -0.020644 -0.00930 -0.07209 -0.01588 0.05646 0.13350 0.21221 0.305047 0.42219 0.42219 0.63471 0.7243 0.744030 -0.07243 -0.07253 -0.01650

577, 86400 76 0073 233, 3844 8 5760 0 0074 - 0 0083 - 0 0	444 -0.09664 -0.485957 908 -0.09664 -0.385957 908 -0.096889 -0.385318 908 -0.01387 -0.2854443 908 -0.096889 -0.385318 909 -0.01387 -0.2854443 909 -0.01387 -0.2854443 909 -0.01387 -0.2854443 909 -0.11213 -0.120333 900 -0.11213 -0.1203531 900 -0.132037 -0.201933 900 -0.132037 -0.201933 900 -0.132037 -0.004501 900 -0.132037 -0.004501 900 -0.1236413 -0.004504 900 -0.1236413 -0.004504 900 -0.1236413 -0.004504 900 -0.123641 -0.004504 900 -0.123651 -0.004504 900 -0.12365 -0.004504 900 -0.12365 -0.004504 900 -0.12365 -0.004504 900 -0.12365 -0.004504 900 -0.12365 -0.004504 900 -0.1555500 900 -0.1555000 900 -0.1555000 900 -0.1555500 900 -0.1555000 900 -0.155000 900 -0.1555000 900 -0	000100211688460849244899305663844139943797789441393875667587644133567884637574783378887603101017848413944567603101017847898764413944567603101017847898787898787898787898789878987898789
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	10	7	4074	EΥ	EY	F7	CP1	CDO	רשא	CP4
550.1262	48.3762 50.4717	Z 287.8208	AREA 134.2319 92.8888	EX 0.1717 0.2388	$0.0014 \\ 0.0017$	0.9852 0.9711	-0.30816	-0.25794 -0.22041	-0.43759 -0.45447	-0.25431 -0.21134
561.9843 571.7032	52.6135	285.3661 282.6921	61.4416	0.2966	0.0010 0.0018	0.9550 0.9307	-0.27718 -0.22829	-0.16922	-0.42744 -0.43130	-0.15505 -0.11874
579.4788 585.5487	54.5321 56.2582	279.9913 277.3716	39.9484 25.6020	0.3658 0.4331	0.0016	0.9013	-0.19732 -0.13264	-0.07718 -0.01227 0.02374	-0.39871	-0.05570 0.00958
590.2199 593.7944	57.8192 59.1648	274.9465 272.7315	16.5738 10.8728 7.3353	0.4939 0.5654	0.0014 0.0029	0.8695 0.8248 0.7812	-0.06057 -0.01973 0.05112	0.02374	-0.37693	0.04253
596.5238 598.6110	60.3196 61.3148		5.0504	0.6243 0.6919 0.7372	0.0023 0.0036	0 7220	0.09611	0.08589 0.12892	-0.36090 -0.37693 -0.32422 -0.32833	0.13801
600.2133 601.4419	62.1788 62.9275	268.9070 267.2740 265.7932	3.5747 2.5984	0.7372 0.8033	0.0033 0.0049	0.6757 0.5956	0.16866 0.18386	0.19697 0.21530	-0.24312 -0.24466	0.19862 0.20207
602.3665 603.0811	64 1591	264.4283 263.1778	1.9398	0.852 <u>5</u> 0.8837	0.0038 0.0027	0.5227 0.4680	0.22415 0.28305	0.25664 0.31624	-0.18414 -0.08194	0.22293 0.2519 <u>3</u>
603.6299 604.0213	64.6628 65.1140 65.5210	262.0277 260.9594	1.4914 1.1617 0.9255	0.9209 0.9558	0.0027 0.0022	0.3899	0.22415 0.28305 0.32533 0.39728	0.36132	-0.00659 0.08741	0.24847 0.25811 0.32924
604.2849 604.4611	65.5210 65.8870	259.9655 259.0447	0.7513 0.6206	0.9764 0.9875	0.0010	0.2158 0.1578	0.53097 0.67215 0.82711	0.54998 0.67378	0.24199	0.46290
604.5779	66.2162	258.1926	V E103	0.9938 0.9970	-0.0007 -0.0012	0.1114	0.82711	0.80740 0.84224	0.60234 0.52489	0.65059 0.79777
604.6533 549.1625	66.5128 57.9387	257.4041 287.9753 285.5306	122.0834	0.1676 0.2341 0.2933	0.0010 0.0013	0.0768 0.9859 0.9722	-0.32713	-0 27132	-0.40972 -0.42294	-0.26745 -0.22039
561.2583 571.1566	58.3482 58.9749	282 8549	0.4391 122.0834 85.1802 56.7892 24.0399 15.3591	0.2933 0.3618	0.0007 0.0014	0.9560 0.9323	0.86056 -0.32713 -0.29405 -0.24406 -0.20676	-0.22981 -0.17823 -0.14238	-0.40203 -0.40751	-0.16400 -0.12453
579.0555 585.2151	58, 9749 56, 39768 61, 19168 62, 23169 62, 23169 63, 32169 64, 32968 64, 32968 64, 32968 65, 59763	280.1478 277.5247 275.0922 272.8690	24:0399	0.4288 0.4907	0.0011	0.9034	-0.14269 -0.06667	-0.08658	-0.38821	-0.06639 -0.00141
589.9539 593.5798	61.1762 61.9198	275.0922 272.8690	10:3591	0.5612	0.0010 0.0024	0.8713	-0 0144R	-0.02097 0.02424	-0.38821 -0.35593 -0.36479 -0.30740 -0.31493 -0.22719	0.04021 0.10511
596.3520 598.4742	62.6074 63.2368	269.0286	4.8777	0.6200 0. <u>6882</u>	0.0017 0.0031	0.7846 0.7255 0.6790	0.06258 0.10620 0.17658	0.09311	-0.31493	0.14288 0.20251
600.1052 601.3595	63.8101 64.3261	267.3842 265.8927	3.4720 2.5388	0.7342	0.0028 0.0042	0.6790 0.6019 0.5275	0.17658	0.20344 0.21984	-0.22022	0.20352
602.3069 603.0382	64.7965 65.2187	264.5167 263.2527	1.9067 1.4731	0.8496 0.8819	0.0033	0.4715	0.18872 0.23740 0.29732 0.33462	0.26948 0.33069	-0.15071 -0.04540	0.23267 0.26109
603.6010	65.5975 65.9463	262.0896 261.0084 260.0010	1.1529	0.9184 0.9543	0.0023 0.0019	0.3955 0.2988	0.40933	0.37112 0.44180	0.02691 0.11916 0.27810	0.24885 0.25771
604.0043 604.2762 604.4575	65.9463 66.2704 66.5685 66.8410 67.0897	260.0010 259.0673	0.9235 0.7529 0.6243	0.9543 0.9757 0.9872	0.0008	0.2191 0.1595	0.55489	0.57051 0.69115	0.37500	0.34858 0.49454
604.5771 604.6539	66.8410	258.2034 257.4052	0.5240 0.4433	0.9937 0.9970	-0.0007 -0.0012	0.1124 0.0770	0.82026	0.79806	0.57941	0.67 <u>3</u> 84 0.80798
548.3271	66.2302 65.2812	288.1093 285.6755	101.4025 72.1869 49.0474	0.1641	0.0008 0.0008	0 9864	-0.33290 -0.30378	0.80640 -0.26557 -0.22961	-0.41139 -0.43299	-0.25916 -0.21966 -0.17218
560.6194 570.6683	64.6583	<b>283 UUUE</b>	49.0474	0.2300 0.2904 0.3581	0.0004 0.0010	0.9732 0.9569 0.9337 0.9053	-0.332590 -0.332378 -0.26033 -0.22280 -0.16277 -0.08859 -0.03254	-0.18559 -0.15216	-0.42418 -0.43197	-0.17218 -0.13554
578.6720 584.9090	64.2951 64.1891 64.2940	277.6653	32.6886 21.5009 14.2292	0.4248 0.4877	0.0007 0.0007	0.9053 0.8730	-0.16277	-0.10403 -0.04330	-0.42039 -0.39329	-0.08572 -0.02634
589.7068 593.3785	64.5048	280.2898 277.6653 275.2277 275.9982	9.5331 6.5530	0.5571 0.6158	0.0019 0.0012	0.8304 0.7879	-0.03254 0.04182	0.00470 0.07265		0.01743 0.08150
596.1893 598.3438	64.5048 64.7733 65.0707	270.9762 269.1446 267.4900 264.9890	4.5956 3.3025	0.6847 0.7312	0.0026 0.0024	0.7288 0.6821	0.08263	0.11480 0.18189	-0.35362 -0.26385	0.11891 0.18022
600.0013 601.2799	65.3774 65.6780 65.9737	265.9890 265.9890	2.4374 1.8469	0.7939 0.8467	0.0035 0.0028	0.6080 0.5321	0.16226	0.19725 0.24157	-0.34527 -0.35362 -0.26385 -0.26156 -0.18813	0.18139 0.20585
602.2491 602.9963		264.6025 263.3258 262.1503 262.0566	1.4375	0.8801 0.9161	0.0020	0.4749 0.4009	0.16226 0.20382 0.25723 0.28599 0.35240	0.29823 0.33214	-0.07948 -0.00484	0.22998 0.20894
603.5725 603.9876	66.5144 66.7661 67.0113 67.2446 67.4651	261.0566 260.0361	1.1330 0.9143 0.7501	0.9529 0.9750	0.0016 0.0007	0.3034	0.35240 0.49909	0.39613 0.52556	0 08131	0.20497 0.29256
604.2678 604.4539	67.2446	259.0898 258.2141	0.6255 0.5274	0.9869 0.9935	0.0006 -0.0006	0.1611 0.1135	0 64007	0.64612	0.23559 0.31791 0.48544	0.44194 0.63134
604.5762 604.6545	67.6657 72.8680 70.9868	257.4063		n 997n	-0.0012 0.0002	0.0773	0.76213 0.76213 -0.32434 -0.29988 -0.25869 -0.21714 -0.15795	() (4h44	0.48544 0.32256 -0.40556	0.77847
547.6589 560.0940	70:9868	288.2168 285.7950	57.7679	0.1613 0.2266 0.2879	0.0004	0.9869 0.9740 0.9577	-0.29988 -0.25869	-0.25553 -0.22533 -0.18284	-0.42394 -0.41479	-0.24643 -0.21493 -0.16980
570.2559 578.3402	69.4597 68.3111	283.1236 280.4127 277.7897	27.7596	0.3549	0.0007 0.0003	0.9349 0.9069	-0.21714	-0.18284 -0.14592 -0.10129	-0.41924	-0.12945 -0.08296
584.6384 589.4844	68.3111 67.5439 67.1006	275.3496 273.1163	78.72 78.7579 40.5140 27.75942 12.6523 8.6328 6.2674	0.4849 0.5534	0.0004	0.8746 0.8329	-0.08434 -0.02338	-0.04409 0.00759	-0.39274 -0.39472 -0.34330 -0.35394 -0.26653 -0.1945	-0.02768 0.01870
593.1944 596.0385	66.8687 66.7807	271.0910	6.0267	0.6120	0.0007 0.0021	0.7908 0.7319	0.05003 0.08820	0.07692 0.11842	-0.34330 -0.35394	0.08402 0.12199
598.2215 599.9030	66.7900 66.8613	269.2534 267.5901	4.2864 3.1161	0.6814 0.7284 0.7895	0.0020 0.0028	0.6851 0.6137	0.15397 0.15987	0.18299 0.19471	-0.26653 -0.26133	0.18290 0.18266
601.2040 602.1934	66.9692 67.1066	266.0809 264.6852	1:7799	0.8439 0.8783	0.0024 0.0017	0.5365 0.4781	0.19049 0.23274	0.22970 0.27755	-0.19452 -0.08589	0.19925 0.21493
602.9558 603.5449	67.2560 67.4081	263.3967 262.2094	1.3970	0.9138 0.9515	0.0014 0.0013	0.4061 0.3078	0.24488 0.29057	0.29844 0.34485	-0.01398 0.05808	0.17822 0.14924
603.9712 604.2593	67.5695 67.7411 67.9138	261.1039 260.0707	0.9030 0.7459 0.6259	0.9743 0.9867	0.0005	0.2252 0.1626	0.43263 0.57796	0.46645 0.58482	0.20095 0.25949	0.22174 0.37499
604.4503 604.5754	68.0813	259.1119 258.2247 257.4073	0.5305	0.9934 0.9970	-0.0008 -0.0013	0.1143 0.0775	0.69351 0.69358	0.67638 0.67524	0.38086	0.57775 0.73803
604.6552 547.1559	68.2406 77.8688	288.2980	0.4510 57.5781	0.1592	-0.0001	0.9872		-0.22339 -0.20249 -0.16702	-0.36594	-0.21365 -0.19246 -0.15420
547.1559 559.6808 569.9186	77.8688 75.4774 73.3882 71.7083	283.2244	32.8694	0.1592 0.2239 0.2859 0.3522 0.4181 0.4825	0.0001 -0.0001 0.0004	0.9872 0.9746 0.9583 0.9359	-0.29037 -0.27487 -0.24232 -0.20209	-0.16702 -0.13143	-0.39491	-0.15420
578.0596 584.403 <u>1</u>	70.4621	277.8981	16.2484	0.4181	0.0000	0.9084 0.8759 0.8352 0.7936 0.7838	-0.14653	-0.09236 -0.04057	-0.39766	-0.11470 -0.07353 -0.02439
589.2865 593.0275 595.8996	70.4621 69.4621 69.0137 68.6313 68.3960 68.2629	273.2235	7.8310	0.5500	0.0010	0.8352	-0.01388 0.05514	0.01194	-0.38827	A A2101
390.10/3	68.3960	269.3550	4:0105	0.6085 0.6783 0.7258	0.0003 0.0017	0.7348 0.6879		0.11867	-0.35906	0.12280
599.8101 601.1315	68.2021 68.2001	266.1686	57.578 32.8694 23.32484 11.2485 7.83178 4.0107 2.2248 1.7200 1.7201	0.7853 0.8412	0.0016 0.0021	0.6191 0.5407	0.14696	0.18602	-0.27237	0.021567 0.08567 0.182280 0.182250 0.18141 0.18468 0.18208
601.1315 602.9165 603.5179 603.9552 604.2510	68.2001 68.1955 68.2267 68.2789 68.3569	263.4653	1.7200	0.8766	0.0019 0.0014 0.0010		0.14815 0.14696 0.15456 0.16781	0.178507 0.178617 0.18602 0.20340 0.22762 0.21964 0.22865 0.36488	0.165694 -0.39443 -0.399455 -0.399566 -0.38827 -0.34393 -0.27814 -0.27208 -0.217814 -0.22008 -0.1175	0.18208
603.5179 603.9552	68.3569	261.1501	1.0896 0.8934 0.7427	0.9116 0.9501 0.9736 0.9864 0.9933 0.9970	0.0010	0.4812 0.4112 0.3120 0.2282 0.1642 0.1153	0.14606 0.14654 0.27051	0.22865		0.11801 0.04152 0.06822
604.2510 604.4468 604.5747	68.5762	259.1339 259.1339	0.7427 0.6268 0.5338	0.9864	0.0005 -0.0001	0.1642	0.43205 0.57230	0.46488	Ŏ.17463	0.20690
	68.6959	288 - 8893 285 - 82444 287 - 5168 287 - 4583 287 - 4583 277 -	0.4548	0.9933 0.9970 0.1577	-0.0006 -0.0011	0.0778	0 60010	0.33652 0.46488 0.57188 0.58984 -0.22331 -0.19537 -0.15538 -0.07825	0.06751	0.43687 0.63529 -0.21195 -0.18476
546.7924 546.7924 559.6447 577.8220 584.1975	81.4904 78.9443 76.5811 74.5840 73.0125	285.3572 285.9623	41.1406 34.5630		-0.0004 -0.0002	0.9875 0.9751	-0.26507	-0.19537	-0.39100	-0.18476 -0.14082
569.6447 577.8220	76.5811 74.5840	283.3065 280.6052	26.8454 19.9052 14.3027	0.2219 0.2842 0.3499 0.4154	-0.0003 0.0001	0.9751 0.9588 0.9368 0.9096	-0.18246	-0.11513	-0.37539	-0.09743 -0.05839
584.1975 589.1091	73.0125 71.8383	280.6052 277.9929 275.5557	14.3027 10.1458	0.4154	-0.0003 -0.0001	0.9096	-0.28746 -0.26507 -0.22679 -0.18246 -0.12600 -0.05479	-0.07861 -0.02925	0.00365 0.13358 0.17463 0.27938 0.06751 -0.37975 -0.379100 -0.37539 -0.37539 -0.37539	-0.01293

\$92.8748 \$70.93497 \$97.9999 \$9.99540 \$9	726676432325288830857777193339823776945144988773262326665321222222222222222222222222222	826068887168795278135648997453351110617300739974622506520857364302311196940839963347168789527813564899745535111061730073997396333471432885736430231119694033967842558827094423232941654842346339558997426652041455827798739873178428843463394673987654458737598739799706308765443463394755665204308765445873724211578099688423776544545737242115780996884237765445457372421157809968842377654458737242115780996884237765467654676767676767676767676767676767	0.5469 0.0007 0.6051 -0.0007 0.67233 0.0012 0.7233 0.0015 0.87812 0.0015 0.88750 0.0001 0.97812 0.0007 0.97832 0.0007 0.97832 0.0007 0.97832 0.0007 0.9932 0.0005 0.1566 0.0007 0.28288 -0.0005 0.34782 0.0005 0.34782 0.0005 0.5440 0.0006 0.5440 0.0006 0.5440 0.0006 0.5440 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.5460 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.98731 0.0007 0.99723 0.0007 0.99724 0.0007 0.99724 0.0007 0.99724 0.0007 0.99724 0.0007 0.99724 0.0007 0.99724 0.0007 0.99724	0.73761 0.07563 0.07563 0.07565 0.06243 0.07565 0.062447 0.03623 0.045492 0.02364 0.02366 0.02	14 0 0 1 1 3 1 1 2 1 2 1 1 1 1 2 1 2 1 1 1 1 2 1 2	3051488488095498236447664372445608845817744639888041966363630131106944556363688612773938360817796888723265324662226537000105555544969688477246580726090833373232140000791000007910000007910000000791000000079100000000	103355593414207375721588708862038822234923844334945693446103775931000000000000000000000000000000000000
003.4140 (1.0134	262.4876 261.3291 260.2377 259.2202 258.2771 257.4124		0.9030 -0.0005		-0.02386 -0.18720 -0.27291	-0.23985 -0.27091 -0.21593 -0.21583 -0.06943 -0.27384	-0.03595

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587.0250 590.2146 592.8365 595.0122 596.8325	79.0599 77.8627 76.8177 75.9080 75.1158	256.9989 256.9989 256.9989 256.9989 256.9989	14.2204 10.5821 7.8709 5.8430 4.3124	0.3420 0.3628 0.3791 0.3936 0.4054	0.9397 0.9319 0.9253 0.9193 0.9141	0.0000 0.0000 0.0000 0.0000 0.0000	-0.02192 0.00111 0.02549 0.05295 0.08433 0.11920	0.03486 0.05567 0.07463 0.09549 0.12026 0.14872	-0.25760 -0.23330 -0.21495 -0.19976 -0.18785 -0.17764	0.05459 0.07226 0.08673 0.10200 0.12086
598.3693 599.6789 600.7998 601.7615 602.5908 603.3087 603.9284	74.4264 73.8231 73.2926 72.8284 72.4227 72.0675 71.7573	256.9989 256.9989 256.9989 256.9989 256.9989 256.9989	3.1615 2.2893 1.6142 1.0987 0.7105 0.4189 0.2039	0.4138 0.4239 0.4322 0.4376 0.4417 0.4455 0.4501	0.9104 0.9057 0.9018 0.8992 0.8972 0.8953 0.8930	0.0000 0.0000 0.0000 -0.0001 -0.0001 0.0000	0.15598 0.19860 0.24514 0.29628 0.35611	0.17928 0.21499 0.25384 0.29612 0.34444 0.39282	-0.17086 -0.16436 -0.16199 -0.15713 -0.15784 -0.14787	0.17082 0.20259 0.23642 0.27223 0.30895 0.34405
604.4529 545.3541 556.2711 565.2228 572.5732 578.35554 583.3950	71.4912 87.8872 86.7467 85.1944 83.5096 81.8553 80.3245	256.9990 262.8315 262.6050 262.2880 261.5443 261.1558	70.2679 70.2679 55.5678 43.6839 33.6554 219.0431	0.4552 0.0702 0.1448 0.2020 0.2492 0.2876 0.3186	0.8904 0.9975 0.9895 0.9794 0.9685 0.9578 0.9479	0.0000 0.0004 0.0005 0.0004 0.0002 0.0000 -0.0001	0.41470 0.51747 -0.27936 -0.31189 -0.25072 -0.16397 -0.09412	0.46130 -0.22696 -0.25391 -0.19843 -0.11821 -0.04874	-0.17319 -1.15990 -1.37832 -1.07777 -0.68844 -0.44143 -0.32944 -0.28283	0.34939 -0.05735 -0.04355 -0.01971 0.00763 0.03376
587 .3100 590 .4912 593 .0965 595 .2503 597 .0448 598 .5540	80.3245 78.9561 77.7545 76.7101 75.0204 74.3413	260.7709 260.3956 260.0315 259.6793 259.3405 259.0132	14.1717 10.5287 7.8242 5.80924 4.21549 2.2947	0.3437 0.3650 0.3807 0.3951 0.4076 0.4139 0.4229	0.9391 0.9310 0.9247 0.9186 0.9132 0.9103 0.9062	-0.0002 -0.0001 0.0004 0.0001 0.0005 0.0010 -0.0017	-0.04925 -0.02101 -0.00005 0.02369 0.05065 0.08014 0.11307 0.14640	0.00296 0.03674 0.05800 0.07625 0.09570 0.11783 0.14370 0.17024	-0.28283 -0.25858 -0.23987 -0.22368 -0.219687 -0.18803	0.05641 0.07364 0.08710 0.10080 0.11689 0.13710 0.15846
599.8361 600.9291 601.8632 602.6632 603.3556 603.9535 604.4606	74.3413 73.7513 73.2339 72.7815 72.3869 72.0431 71.74873	258.6970 258.3934 258.1028 257.8250 257.5623 257.5188 257.1007	1.6261 1.1137 0.7258 0.4318 0.2124 0.0567	0.4322 0.4389 0.4441 0.4465 0.4503 0.4554	0.9018 0.8985 0.8960 0.8948 0.8929 0.8903	-0.0042 -0.0036 -0.0001 0.0034 0.0030	0.18292 0.22377 0.27308 0.33413 0.41014 0.49638	0.19930 0.23141 0.26980 0.31763 0.37831 0.44826	-0.17809 -0.17140 -0.16217 -0.15711 -0.15006 -0.14933	0.18118 0.20487 0.23191 0.26422 0.30595 0.31090
545.4229 556.5296 565.7042 573.2260 579.3151 584.2049	87.8788 86.7038 85.0908 83.3376 81.6233 80.0483	268.6531 268.1743 267.5102 266.7581 265.9680 265.1722	70.8902 56.5243 44.2912 33.8806 25.4233 18.8611	0.0708 0.1464 0.2052 0.2535 0.2926 0.3240	0.9975 0.9892 0.9787 0.9673 0.9562 0.9461 0.9371	0.0008 0.0011 0.0009 0.0006 0.0005 0.0004	-0.22580 -0.25899 -0.20176 -0.13338 -0.08165 -0.04429 -0.01828	-0.19955 -0.22332 -0.14984 -0.06048 -0.00473 0.02506 0.04732	-1.11931 -1.31766 -0.67634 -0.44806 -0.34431 -0.27665 -0.257216	-0.06160 -0.04701 -0.02538 0.02509 0.06579 0.07753 0.07959
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601.3089 602.1627 602.8845 603.5025 604.0343 604.4856 545.5417	73.5387 73.0567 72.62780 71.7025 71.7025 71.8643	259.7264 259.1668 258.6298 258.1182 257.6391	1.6398 1.1409 0.7587 0.4619 0.2330 0.0638 71.8002 57.5730	0.4340 0.4419 0.4477 0.4472 0.4495 0.4547 0.0718	0.9009 0.8970 0.8942 0.8944 0.8933 0.8906 0.9974	-0.0032 -0.0048 -0.0033 0.0015 0.0023 -0.0001 0.0013	0.15832 0.19024 0.23549 0.30362 0.39125 0.53572 -0.20806 -0.22643	0.17863 0.20121 0.23369 0.28416 0.35395 0.46051 -0.19779	-0.19487 -0.18075 -0.16889 -0.15406 -0.13232 -0.18499 -0.98034	0.16941 0.16941 0.18779 0.21802 0.27299 0.32078
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460.9447 474.5657 489.6806 505.6255	79.5896 79.8019 80.0691 80.3970 80.7629 81.0719 77.9421 78.0476 78.1579 78.4045	225.3959 224.08987 222.4920 222.920 222.920 224.0329 256.6164 254.9453 254.9453 252.9841	32.8710 0.0023	0.0000 0.0001 0.0006 0.0006 -0.0025 0.0003 0.0003 0.0002	-0.9855 -0.9938 -0.9985 -1.0000	-0.140834 -0.30938 -0.23531 -0.23531 -0.22530 -0.181994 0.97687 0.98149 1.04829 1.06259	-U 43063	n 1 n 9 6 9	-0.59688 -0.29214 -0.19887 -0.18930 -0.15377 1.15183 1.09606 1.095048 1.02013

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431.9033	84.2672	233.5024	4.3425	-0.3526	0.0000	-0.9358	-0.35402	-0.22230	0.41800	-0.64469
439.7852	84.5426	230.8612	5.3322	-0.2891		-0.9573	-0.46614	-0.39203	0.31698	-0.78338
449.5545	84.8790	228.2767	6.5311	-0.2279	-0.0001	-0.9737	-0.47950	-0.47906	0.20889	-0.79956
461.3170	85.2812	225.9322	7.8740	-0.1676	-0.0001	-0.9858	-0.40507	-0.42808	0.10150	-0.58769
474.9622	85.7480	224.0459	9.1723	-0.1096	-0.0001	-0.9940	-0.30703	-0.29480	0.00584	-0.28540
490.1018	86.2683	222.8277	10.1818	-0.0529	0.0002	-0.9986	-0.25077	-0.20459	-0.08472	-0.19587
506.0705	86.8192	222.4420	10.6623	0.0039		-1.0000	-0.23358	-0.18813	-0.18706	-0.18752
522.0361 534.6664	87.3718 87.8086	222.9518 223.9528	10.5329	0.0606	0.0005 -0.0027	-0.9982 -0.9940	-0.22520 -0.19222	-0.19229 -0.17107	-0.31280 -0.43914	-0.19211 -0.15685

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401.765.66.66.66.66.66.66.66.66.66.66.66.66.6	07853307333084411399994339937668128429443993766908888208809977713998872689990123333033333333333333333333333333333	711444420917 0457831123833075557199011277422986556717744830681157766800330385622062442110220775311443099156303317524511409117093555211443091156901138442110411141509315651138442110411415093141509317527841140911704551131443091111111111111111111111111111111	09999999999999999999999999999999999999	44444444444444444444444444444444444444	0.000000000000000000000000000000000000	-0.11868 -0.12894 -0.138949 -0.14615505 -0.1215505 -0.1215505 -0.1215505 -0.1215505 -0.1215505 -0.1215506 -0.1215506 -0.121506 -0.1	935777179933168 935777177933168 935777177933168 9357798779857987798798798798798798798798798798798798	-0.098745 -0.12110 -0.137965 -0.159078 -0.2282166 -0.2282166 -0.2282641 -0.278641 -0.30028 -0.313587 -0.11628818 -0.0167548 -0.0466183 -0.0466183 -0.0466183	0.1258
400.2922 83.4542 401.1964 83.4857 401.9385 83.517 402.6247 83.5357 403.3594 83.5613 404.1411 83.5886 404.9788 83.6179 406.8873 83.6845	301.0378 301.4090 301.6179 301.8627 302.2257 302.6721 303.7169 304.2938	16.0631 10.9915 7.8670 5.8549 4.5703 3.8305 3.1275 3.4922 3.7842 4.0581 4.3528 4.7026 5.1277	0.0349 0.0349 0.0359 0.0349 0.0349 0.0349 0.0349	-0.9994 -0.9994 -0.9994 -0.9994 -0.9994 -0.9994 -0.9994	0.0000 0.0000 0.0000 0.0000 0.0000 0.0000	0 -0.10596 0 -0.11393 0 -0.12310 0 -0.13270	0.01936 -0.00198 -0.01064 -0.03703 -0.05102 -0.06606 -0.08177	-0.04619 -0.05018 -0.05613 -0.06489 -0.07359	0.12587 0.11806 0.11102 0.09832 0.08286 0.06762

40890143860423210825031882102937796612713556686 027349391493860423213853548833548299377966127124608666686868686868686868686868686868686	8390992315588196925448849996777614920225586600220005027761859860196992315986647776149207235586002200050277688990317915888333259986443670030030333554667149207788899331591414920778889993159141492077888999315914149207788898993159141492077888989931591414920778889899315914149207888888888888888888888888888888888888	304.5274833 305.6.5274833 305.6.5274853 307.8.86532 307.86532 30	746491271656128950718007126590314336846253149915699318444550142995071800712465903143368462005668921695130080991513068462207663349833755109597966892169513008099910706991471155374607769114770481233203809910706913471153741226591147711537320332328096117543233333328096117543233333332809611754323333333328096117543233333333333333333333333333333333333	034494999999999999999999999999999999999	$\begin{array}{c} -0.\\ -0.\\ -0.\\ -0.\\ -0.\\ -0.\\ -0.\\ -0.\\$	0.0000 0.00000	-0.15469777449667766169151441912224632544796857769952369595836655882749667766616776661677676676767676767676767	-00-01-13-14-94-14-03-15-92-98-7-7-09-14-00-17-18-7-12-15-15-98-16-15-94-12-17-18-15-15-15-15-15-15-15-15-15-15-15-15-15-	-0.100000000000000000000000000000000000	96788319740754000000000000000000000000000000000
442.0945	84.9140	315.1545	31.1679	0.0349	-0.9994	0.0000	-0.20045	-0.22911	-0.14353	-0.22618
451.4866	85.2420	315.2244	35.2051	0.0349	-0.9994		-0.21328	-0.21872	-0.17491	-0.22823

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406.4647 407.0364 407.8042 408.8045 410.1005	77.2786 77.1298 76.9664 76.7848 76.5813	262.1399 263.3269 264.6108 266.0098 267.5354	2.6583 -0.9212 3.0691 -0.8810 3.6346 -0.8368 4.3342 -0.7920 5.2982 -0.7354	0.0004 0.0004 0.0003 0.0005 0.0004	0.3891 0.4312 0.4730 0.2800 0.5475 0.1424 0.6106 0.0111 0.6777 -0.1421 0.7330 -0.2680	6 0.32127 0 0.19825 5 0.08152 8 -0.05744	-0.16386 -0.26869 -0.33323 -0.38264 -0.45335 -0.47716	0.80769 0.69346 0.57187 0.44699 0.30068
413.76243 413.96243 416.8026 420.4928 425.2838 431.4602	76.3510 76.0904 75.7976 75.4704 75.1282 74.7919	269.2114 271.0574 273.0876 275.3166 277.7445 280.3284	6.6152 -0.6802 8.4244 -0.6164 10.9375 -0.5527 14.4492 -0.4876 19.3067 -0.4230 25.7950 -0.3557	0.0005 0.0004 0.0003 0.0004 0.0007	0.7874 -0.3954 0.8334 -0.4917 0.8731 -0.5628 0.9061 -0.6002 0.9346 -0.6085	0 -0.29792 6 -0.40148 1 -0.49179 0 -0.56167 3 -0.61333	-0.51391 -0.51444 -0.51133 -0.49211 -0.47880	0.02073 -0.11263 -0.23890 -0.35114 -0.44910
439.3020 449.0454 460.8095 474.5019 489.7407 505.8351	74.4697 74.2049 74.0797 74.2216 74.7611 75.7367 77.0872	282.9863 285.5954 285.9783 289.9056 291.1457 291.5508	34.2474 -0.2925 44.7195 -0.2304 56.5690 -0.1712 68.0679 -0.1112 76.4482 -0.0533 79.0980 0.0023	-0.0001 0.0000 0.0007 0.0009 0.0003 0.0000	0.9563 -0.5783 0.9731 -0.5325 0.9852 -0.4859 0.9938 -0.4659 0.9986 -0.4494 1.0000 -0.4304	7 -0.53923 5 -0.45087 8 -0.40293 0 -0.38812	-0.45266 -0.43935 -0.42711 -0.42697 -0.42455 -0.41959	-0.50952 -0.51832 -0.47967 -0.43232 -0.40106 -0.38023
521.9149 535.0737 405.3670 405.4257 405.5424 405.7427	78.4790 79.6087 79.5455 79.4769 79.4029	291.0605 289.9954 257.3910 258.2049 259.0852 260.0354	75.1858 0.0594 47.6698 0.1127 1.2578 -0.9990 1.3899 -0.9953 1.5446 -0.9867 1.7302 -0.9693	0.0003 -0.0009 0.0003 0.0003 0.0003	0.9982 -0.4076 0.9936 -0.3656 0.0445 0.9364 0.0972 0.8084 0.1625 0.7609 0.2459 0.6247	1 -0.30819 1 0.92789 5 0.81792 1 0.77246 0.65157	-0.43011 -0.41467 0.35374 0.20058 0.18659 0.04915	-0.35486 -0.30272 1.19443 1.08895 1.06317 0.96085
406.0499 406.4765 407.0543 407.8296 408.8390 410.1462	79.3234 79.2369 79.1434 79.0427 78.9328 78.8123	261.0592 262.1658 263.3587 264.6483 266.0529 267.5837 269.2645	1.9535 -0.9459 2.2315 -0.9205 2.5722 -0.8798 3.0399 -0.8357 4.6174 -0.7903 4.4103 -0.7337	0.0003 0.0003 0.0003 0.0003 0.0004 0.0003	0.3246 0.5192 0.3907 0.4004 0.4753 0.2537 0.5492 0.1214 0.6127 -0.0072 0.6795 -0.1568	0.44681 0.31191 0.19011 0.07265 -0.06604	-0.05927 -0.16468 -0.2723 -0.33897 -0.46275 -0.49103 -0.53269	0.90293 0.81196 0.69047 0.56628 0.44132 0.29616
411.8411 414.0389 416.8948 420.6047 425.4156 431.6092	78.6791 78.5329 78.3749 78.2074 78.0483 77.9179	269.2645 271.1147 273.1477 275.3780 277.8038 280.3838	5.4896 -0.6784 6.9640 -0.6146 8.9994 -0.5509 11.8213 -0.4859 15.6761 -0.4212 20.7600 -0.3546	0.0004 0.0003 0.0002 0.0003 0.0006 0.0003	0.7347 -0.2810 0.7888 -0.4070 0.8346 -0.5026 0.8740 -0.5725 0.9070 -0.6090 0.9350 -0.6151	-0.18384 -0.30814 7 -0.41303 1 -0.50344 5 -0.57328	-0.49103 -0.53269 -0.53534 -0.52892 -0.50674 -0.49029	0.16156 0.01298 -0.12245 -0.24959 -0.36265 -0.45990
439.4661 449.2209 460.9881 474.6682 489.8763 505.9288	77.8289 77.8176 77.9455 78.2986 78.9554 79.9211	283.0367 285.6371 288.0066 289.9207 291.1517 291.5505	27.2684 -0.2914 35.1255 -0.2293 43.6964 -0.1700 51.5300 -0.1107 56.5545 -0.0529 57.0556 0.0026	-0.0001 -0.0006 0.0009 0.0003 0.0000	0.9565 -0.5835 0.9734 -0.5369 0.9854 -0.4928 0.9939 -0.4677 0.9986 -0.4505 1.0000 -0.4300	3 -0.61533 1 -0.54544 1 -0.45504 1 -0.40519 9 -0.38983	-0.46416 -0.44881 -0.43397 -0.43008 -0.42601 -0.41887	-0.51972 -0.52655 -0.48534 -0.43512 -0.40268 -0.38005
521.9684 534.8994 405.3674 405.5442 405.7463	81.1277 82.2757 81.0713 81.0328 80.9918 80.9485	291.0562 290.0185 257.3921 258.2093 259.0933 260.0476	52.8183 0.0597 31.4507 0.1117 1.0356 -0.9990 1.1438 -0.9952 1.2703 -0.9866 1.4217 -0.9690	0.0002 -0.0009 0.0003 0.0003 0.0003	0.9982 -0.4032: 0.9937 -0.3524 0.0445 0.8328: 0.0976 0.7313: 0.1632 0.6853: 0.2471 0.5681: 0.3256 0.4695	0.29652 0.86925 0.77433 0.73007 0.62024	-0.42711 -0.40904 0.37814 0.21981 0.20520 0.05408	-0.35144 -0.28971 1.18451 1.07991 1.05357 0.95538
406.0562 406.4862 407.0690 407.8505 408.8674 410.1837	80.9033 80.8555 80.8055 80.7539 80.6999 80.6437	261.0758 262.1872 263.3849 264.6791 266.0883 267.6234	1.6037 -0.9455 1.8300 -0.9199 2.1064 -0.8788 2.4852 -0.8347 2.9519 -0.7890 3.5908 -0.7324	0.0003 0.0003 0.0003 0.0003 0.0003	0.3921 0.3579 0.4772 0.2200 0.5506 0.09593 0.6144 -0.02800 0.6809 -0.1714	0.42163 0.29088 0.17269 0.05645 -0.07939	-0.05900 -0.16138 -0.27125 -0.34140 -0.39974 -0.47309	0.89622 0.80140 0.67593 0.55286 0.42920 0.28508
411.8895 414.0997 416.9698 420.6953 425.5215 431.7281	80.5855 80.5267 80.4708 80.4225 80.3973 80.4131	269.3081 271.1614 273.1966 275.4277 277.8516 280.4280	4.4578 -0.6769 5.6362 -0.6131 7.2540 -0.5494 9.4812 -0.4844 12.4895 -0.4197	0.0003 0.0002 0.0002 0.0002 0.0005 0.0003	0.7361 -0.2926 0.7900 -0.4166 0.8356 -0.5111 0.8748 -0.5791 0.9077 -0.61373 0.9354 -0.6172	5 -0.32009 2 -0.42479 5 -0.51401 3 -0.58217 -0.62954	-0.50256 -0.54359 -0.54514 -0.53636 -0.51337 -0.49544	0.14983 0.00133 -0.13422 -0.26069 -0.37274 -0.46788
439.5957 449.3579 461.1253 471.7936 489.9763 505.9963	80.4835 80.6369 80.9150 81.3706 82.0444 82.9234	283.0766 285.6697 285.9320 289.9320 291.1562 291.5503	21.3434 -0.2905 27.1469 -0.2285 33.2349 -0.1692 38.4200 -0.1103 41.1738 -0.0527 40.4106 0.0028	-0.0002 -0.0002 0.0006 0.0008 0.0003 0.0000	0.9569 -0.5847 0.9736 -0.5365 0.9856 -0.4917 0.9939 -0.4659 0.9986 -0.4487 1.0000 -0.4276	-0.61909 -0.54683 -0.45477 -0.40423 -0.38907 -0.37535	-0.46916 -0.45227 -0.43614 -0.43070 -0.42639 -0.41789	-0.52588 -0.53033 -0.48683 -0.43486 -0.40215 -0.37879
522.0060 534.7821 405.3678 405.4272 405.5457 405.7493	83.9451 84.8592 82.2607 82.2219 82.2026	291.0533 290.0341 257.2931 258.2129 259.0999 260.0575	36.2775 0.0600 20.4351 0.1110 0.8359 -0.9990 0.9228 -0.9952 1.0244 -0.9865 1.1457 -0.9688	0.0002 -0.0009 0.0003 0.0003 0.0003	0.9982 -0.3979 0.9938 -0.3367 0.0446 0.7495 0.0978 0.6851 0.1638 0.6439 0.2480 0.5489	0.73549	-0.42287 -0.39739 0.39910 0.23365 0.21016 0.05234	-0.34742 -0.27530 1.16997 1.06961 1.04782 0.95023
406.4940 406.4940 407.0809 407.8674 408.8903 410.2139	82.1663 82.1500 82.1362 82.1252 82.1183	261.0893 262.2046 263.4060 264.7041 266.1169 267.6553	1.2914 -0.9452 1.4723 -0.9195 1.6927 -0.8780 1.9943 -0.8340 2.3653 -0.7879 2.8718 -0.7313	0.0003 0.0003 0.0002 0.0002 0.0003	0.3932 0.3514 0.4787 0.2199 0.5518 0.0968 0.6157 -0.0285 0.6821 -0.1786 0.7372 -0.2960	0.27940 0.16463 0.04903 0.08663	-0.05852 -0.15786 -0.26957 -0.34006 -0.47850 -0.51030	0.78685 0.66257 0.54338 0.42146 0.27804
411.9285 414.1485 417.0297 420.7673 425.6053 431.8216	82.1171 82.1243 82.1447 82.1842 82.2556 82.3735	269.3430 271.1989 273.2357 275.4672 277.8894 280.4628	3.5574 -0.6757 4.4852 -0.5482 7.4878 -0.4833 9.8082 -0.4833 16.5052 -0.3530 16.5052 -0.2899 20.7574 -0.2278 25.4437 -0.1685	0.0001 0.0001 0.0001 0.0004 0.0002	0.7909 -0.42083 0.8364 -0.5147 0.8754 -0.58178	-0.32811 -0.43220 -0.52028 -0.58702 -0.63273	-0.54983 -0.54817 -0.53678 -0.51382 -0.49608	0.14272 -0.00544 -0.14101 -0.26690 -0.37809 -0.47185 -0.52880 -0.53208
439.6967 449.4633 461.2294 474.8871 490.0492 506.0442 522.0318	82.11447 822.14447 822.14556 822.18556 822.357355 822.85266 832.16665 833.16665 833.16665 834.26456 835.2656 835.2656	277 . 8894 280 . 4628 283 . 1076 285 . 6949 288 . 0450 289 . 9405 291 . 15502 291 . 0513 297 . 0343 258 . 2157 259 . 1054	28.4123 -0.1100 29.7433 -0.0525	-0.0002 0.0005 0.0008 0.0003 0.0000 0.0002	0.9857 -0.48998	-0.45488 -0.40387	-0.47064 -0.45347 -0.43718 -0.43116 -0.42669 -0.41915	-0.48765
534.7053 405.3680 405.4278 405.5468 405.7516 406.0652	85,71613 863,22030 833,22033 833,22033 833,22033 833,22033 833,22033 833,22033 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568 833,233,568	290.0443 257.3938 258.2157 259.1052 260.0654 261.1000	24. 6134 0.0601 13.1733 0.1105 0.6640 -0.9990 0.7328 -0.9952 0.8131 -0.9854 0.9089 -0.9686 1.0238 -0.9450	-0.0009 0.0002 0.0002 0.0003 0.0003	0.9986 -0.4471; 1.0000 -0.44259; 0.9982 -0.3945; 0.9939 -0.3331; 0.0447 0.7422; 0.1642 0.7044; 0.1642 0.5734; 0.3270 0.3665; 0.4799 0.2257; 0.5527 0.9559; 0.6168 -0.0337;	-0.45488 -0.4848866 -0.388866 -0.34491 -0.27869 0.79531 0.79531 0.69037 0.50770	-0.41915 -0.40248 0.41320 0.24231 0.20472 0.04380 -0.06085	-0.43486 -0.40193 -0.37802 -0.34509 -0.2716224 1.06481 1.04812 0.948997 0.77587
406.5003 407.0903 407.8807 408.9084 410.2379	83.2103 83.2198 83.2350 83.2568 83.2872 83.3290	261 . 1000 262 . 2184 263 . 4229 264 . 7239 266 . 1395 267 . 6806 269 . 3707	1.0238 -0.9450 1.1665 -0.9191 1.3398 -0.8773 1.5767 -0.8334 1.8677 -0.7871 2.2643 -0.7304 2.7997 -0.6748 3.5219 -0.6110 4.5051 -0.5472	0.0003 0.0002 0.0002 0.0003 0.0002 0.0002	0.3940 0.36657 0.4799 0.22578 0.5527 0.09598 0.6168 -0.03377 0.6830 -0.18168 0.7380 -0.30442	0.59700 0.59774 0.40530 0.27882 0.16127 0.09474 -0.09474	-0.27037 -0.33988 -0.40233	0.77587 0.65363 0.53783 0.41624 0.27213 0.13570 -0.01332
411.9590 414.1870 417.0769 420.8238 425.6708	83.3857 83.4627 83.5665 83.7071	267.16806 269.3707 271.2285 273.2664 275.4983 277.9190	3.5219 -0.6110 4.5051 -0.5472 5.8431 -0.4824 7.6182 -0.4177	0.0001 0.0000 0.0001 0.0003	0.6830 -0.03316 0.7380 -0.3044 0.7980 -0.42877 0.8370 -0.52073 0.8759 -0.58520 0.9086 -0.61596	-0.09474 -0.09474 -0.21255 -0.33673 -0.43961 -0.52598 -0.59075	-0.51847 -0.51847 -0.55469 -0.53469 -0.531449	-0.01332 -0.14871 -0.27368 -0.38350

431.8942 439.7744 449.5437 461.3077 474.9562 490.1019 506.0494	83.8956 84.1431 84.4612 84.8603 85.3456 85.9121 86.51796	280.4897 283.1315 285.7140 288.0574 289.9468 291.1619 291.5500 291.0500	9.8846 12.6537 15.7588 18.7684 20.9315 21.4291 19.86147	-0.3525 -0.2893 -0.2273 -0.1680 -0.1097 -0.0524 0.0031	0.0002 -0.0003 -0.0002 0.0005 0.0008 0.0002	0.9358 0.9572 0.9738 0.9858 0.9940 0.9986 1.0000	-0.61572 -0.58158 -0.58251 -0.48781 -0.46236 -0.44596 -0.42486	-0.63418 -0.62035 -0.54587 -0.45307 -0.40264 -0.38792 -0.37378	-0.49572 -0.47060 -0.45379 -0.43780 -0.43151 -0.42680 -0.416324	-0.47530 -0.53034 -0.53198 -0.48675 -0.43384 -0.40117 -0.37717
522.0494	87.1796	291.0500	16.6147	0.0602	0.0002	0.9982	-0.39074	-0.34188	-0.41364	-0.34195
534.6558	87.7018	290.0508	8.4641	0.1101	-0.0009	0.9939	-0.32369	-0.26964	-0.39696	-0.26182

70078316780488996675064419506535844557250096535843647678899654906549065490654996413841218099994667503778899966473844233446785967530619350653333333333333333333333333333333333	2675986238200676763930409914483517463763357619904587791167173722413647855511113738506814478836269944699770898244297483842601844438511403376467238142524144639974813248142524111111237384425631444513144033764672881429744833837192465334656384414474413328333817192464788424673441440337646728842467394944674794848441441332843434444033844246744445133144677994944467479494467479494947474634744637464747494747494747474747474747474747474	90095100872018888489080252110997176563299700800748976381421745058377688112112278832401443399124445017759110	\$\frac{\text{REA}}{2}\$ = \frac{\text{PRINTED}{\text{PRINTED}}{2}\$ =	00140444444444444444444444444444444444	699299960000000000000000000000000000000	-0.10713 -0.11877 -0.13704 -0.17053 0.26663	\$\begin{array}{l} \text{1043445} \text{10487} \\ \text{1043445} \\ \text{104345} \\ \text{1043445} \\ \text{104345} \\ \text{1043445} \\ \text{104345} \\ \text	-0.196645 -0.4755789 -0.4755789 -0.4755789 -0.43318937 -0.23315361 -0.228347 -0.228347 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.228348 -0.116548 -0.116548 -0.116548 -0.116848	-0.09602 -0.12059 0.02197
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plate top upper

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390.8024 394.5778 397.3473 399.3694 400.8507	85.6016 85.7578 85.8531 85.9235 85.9753	282.1503 283.4859 284.3304 284.7061	21.9187 -0.0458 17.8391 -0.0342 13.9616 -0.0347 10.8220 -0.0349 8.1238 -0.0349	0.9994 0.9994 0.9994	0.0000	-0.21028 -0.19665 -0.18673	-0.33978 -0.23264 -0.22351	-0.29771 -0.28009 -0.26661	-0.63088 -0.57749 -0.46029

401.8430 402.6134	86.0099 86.0368	284.8721 285.1928 285.7930	4.7747 -0.0348 5.0881 -0.0349 5.3484 -0.0350	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.17842 -0.17062 -0.16300	-0.22072 -0.21053 -0.19879	-0.25725 -0.24931 -0.24072	-0.35731 -0.31867 -0.29313
403.4548 404.3754 405.3890 406.5112	86.0662 86.0984 86.1338 86.1729	286.6256 287.6244 288.7270	5.6154 -0.0349 5.9145 -0.0349 6.2603 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.15613 -0.14997 -0.14459	-0.18862 -0.18035 -0.17372	-0.23335 -0.22586 -0.21935	-0.27247 -0.25561 -0.24196
407.7584 409.1499 410.7086	86.2165 86.2651 86.3195	289.8857 291.0704 292.2654	6.6590 -0.0349 7.1281 -0.0349 7.6737 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.13998 -0.13615 -0.13288	-0.16827 -0.16374 -0.15970	-0.21317 -0.20774 -0.20260	-0.23079 -0.22146 -0.21297
412.4645 414.4592 416.7503	86.3809 86.4505 86.5305	293.4654 294.6721	8.3360 -0.0349 9.1555 -0.0349 10.1905 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.13006 -0.12756 -0.12529	-0.15577 -0.15169 -0.14725	-0.19779 -0.19300 -0.18808	-0.20483 -0.19689 -0.18880
419.4229 422.6084 426.5000	86.6238 86.7351 86.8710	295.8888 297.1175 298.3542 299.5832	11.5407 -0.0349 13.3357 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.12309 -0.12110 -0.11906	-0.14237 -0.13722 -0.13181	-0.18271 -0.17692 -0.17019	-0.18047 -0.17206 -0.16390
431.3790 437.6231 445.5853	87.0413 87.2594 87.5375	300.7614 301.8650 302.9150	18.5738 -0.0349 21.9979 -0.0349 25.3263 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.11699 -0.11452 -0.11209	-0.12597 -0.11914 -0.11103	-0.16274 -0.15449 -0.14598	-0.15724 -0.15096 -0.14522 -0.13495
455.4134 466.9681 479.8416	87.8807 88.2841 88.7338	303.9026 304.7698 305.4486	18.55738 -0.0349 21.9979 -0.0349 25.3263 -0.0349 27.8127 -0.0349 29.2710 -0.0349 29.2710 -0.0349 28.5897 -0.0349 27.3265 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.11002 -0.11030 -0.11075	-0.10136 -0.09317 -0.08548	-0.13751 -0.13236 -0.12854	-0.12240 -0.10811
493.4023 506.7681 518.3573	89.2074 89.6741 90.0787	305.9075 306.1462 306.0228	44.0000 0.0040	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.11951 -0.13315 -0.16674	-0.08664 -0.09329 -0.11771	-0.12937 -0.15846 -0.10475	-0.10172 -0.09968 -0.11027
375.6508 378.9165 382.5759	83.9426 84.5457 84.9936	282.5457 281.7774 282.0323	16.5077 -0.1419 15.9038 -0.1000	0.9752 0.9898 0.99 <u>4</u> 9	0.0191 -0.0101 -0.0123	-0.14669 -0.29309 -0.32456	-0.24713 -0.37379 -0.47988	-0.30230	-0.45196 -0.55140
387.2053 391.9248 395.3233	85.3814 85.6560 85.7838	283.5159 285.8144 287.8926	23.6447 -0.0689 18.2029 -0.0400 14.6130 -0.0337	0.9976 0.9992 0.9994	-0.0043 -0.0006 -0.0001	-0.32456 -0.32701 -0.31838 -0.25978	-0.50649 -0.55343 -0.50344	-0.37010 -0.48923 -0.48927 -0.39768 -0.33552 -0.31332 -0.27679 -0.26352 -0.25232	-0.57601 -0.63467 -0.65177
397.8106 399.6617 401.0715	85.8691 85.9338 85.9830	289.4548 290.4652	11.4526 -0.0350 9.1678 -0.0348 7.1986 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.25978 -0.22490 -0.21463 -0.20329	-0.50344 -0.37641 -0.30181 -0.28532	-0.31332 -0.29286 -0.27679	-0.63429 -0.59941 -0.53483
402.0475 402.8255 403.6859	86.0170 86.0442 86.0743	290.9408 291.1199 291.3785 291.8649	4.3435 -0.0349 4.7989 -0.0349 5.1186 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.18579 -0.17657	-0.26503 -0.24904	-0.26352 -0.25232 -0.24156	-0.47202 -0.42917 -0.39650
404.6210 405.6338 406.7346	86.1069 86.1423 86.1807	291.8649 292.5320 293.3218 294.1862	5.3895 -0.0349 5.6615 -0.0349 5.9686 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.16802 -0.16012 -0.15292 -0.14656	-0.23386 -0.22051 -0.20841	-0.24156 -0.23301 -0.22490 -0.21838	-0.36737 -0.34196 -0.31915
407.9380 409.2644 410.7396	86.2228 86.2691 86.3206	295.0926 296.0240 296.9742	6.3278 -0.0349 6.7731 -0.0349 7.3147 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.14103 -0.13637	-0.19729 -0.18702 -0.17776	-0.21235 -0.20711 -0.20202 -0.19692 -0.19134	-0.29650 -0.27518 -0.25593
412.3978 414.2859 416.4661	86.3785 86.4444 86.5206	297.9440 298.9416 299.9675	8.0051 -0.0349 8.8839 -0.0349 10.0027 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.13254 -0.12945 -0.12704	-0.16956 -0.16236 -0.15578		-0.22377 -0.21064
419.0285 422.1113 425.9093	86.6100 86.7177 86.8504	301.0194 302.0835 303.1294	11.4637 -0.0349 13.3968 -0.0349 15.8370 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.12508 -0.12353 -0.12204 -0.12066	-0.14931 -0.14213 -0.13382 -0.12449	-0.17780 -0.16928 -0.15945	-0.19855 -0.18539 -0.17040 -0.15288
430.7009 436.8526 444.6847	87.0176 87.2325 87.5060	304.0892 304.9332 305.7109	18.8247 -0.0349 22.2761 -0.0349 25.5767 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0000 0.0000 0.0000 0.0000	-0.11945 -0.11876 -0.11910	-0.11450 -0.10627 -0.10027	-0.14870 -0.13806 -0.13149	-0.13654 -0.12356 -0.11585
454.3155 465.6054 478.1623 491.3845	87.8423 88.2366 88.6751	306.4414 307.0892 307.6079	25.5767 -0.0349 28.0695 -0.0349 29.4009 -0.0349 29.5940 -0.0349 28.8233 -0.0349 27.2559 -0.0349	0.9994 0.9994 0.9994	0.0000	-0.12191 -0.12548	-0.09732 -0.09514 -0.10142	-0.12966 -0.13292 -0.13804 -0.14592	-0.11142
504.4695 515.9168 378.6060	89.1369 89.5938 89.9935 84.4609	307.6079 307.9770 308.2251 308.3160 286.8519	27.2559 -0.0349 23.1434 -0.0349 15.7942 -0.1887	0.9994 0.9994 0.9817	0.0000 0.0000 0.0252	-0.13724 -0.15283 -0.18860 -0.24701	-0.11090 -0.13651 -0.37397	-0.17569 -0.13307 -0.64556	-0.10872 -0.11192 -0.12702 -0.50863
381.5224 384.7188 388.8268	84.9206 85.2346 85.4993 85.7001	286.1645 286.6092 288.4181	13.8058 -0.1165 12.7238 -0.0765 19.8177 -0.0526	0.9932 0.9970 0.9986	-0.0066 -0.0104 -0.0036	-0.24701 -0.26188 -0.32107 -0.32480 -0.30304	-0.39451 -0.50137 -0.52059	-0.36173 -0.55636 -0.46155	-0.49404 -0.59206 -0.61502
393.0424 396.0241 398.2182	85.7001 85.8077 85.8833	291.0376 293.2645 294.9059	14.9729 -0.0378 11.7904 -0.0341 9.2132 -0.0350	0.9994 0.9994	0.0004 0.0001 0.0000	-0.30304 -0.26259 -0.24357	-0.53751 -0.48923 -0.41631	-0.43568 -0.39435 -0.37214 -0.34860 -0.32841 -0.30975	-0.65474 -0.65888 -0.63823
399.8861 401.1944 402.1253	85.9416 85.9872 86.0197	295.9733 296.5047 296.7093	7.4329 -0.0348 5.9741 -0.0349 3.7083 -0.0350	0.9994 0.9994 0.9994	0.0000	-0.26259 -0.24357 -0.23511 -0.22538 -0.21869	-0.37930 -0.36286 -0.35277	-0.34860 -0.32841 -0.30975	-0.61202 -0.58076 -0.55795 -0.54165
402.8753 403.6976 404.5806	86.0460 86.0747 86.1055	296.9389 297.3442 297.8808	4.1510 -0.0349 4.4692 -0.0348 4.7412 -0.0349	0.9994 0.9994 0.9994	0.0000	-0.21117 -0.20275 -0.19430	-0.34005 -0.32547 -0.31023	-0.27856 -0.26676	-0.52140 -0.49780
405.5261 406.5452 407.6545	86.1385 86.1741 86.2129	298.5031 299.1789 299.8896	5.0170 -0.0349 5.3347 -0.0349 5.7155 -0.0349	0.9994 0.9994 0.9994		-0.18590 -0.17741 -0.16919	-0.29524 -0.27979 -0.26366	-0.25588 -0.24697 -0.23876	-0.47364 -0.44812 -0.42005
408.8782 410.2462 411.7975	86.2556 86.303 <u>4</u> 86.3575	300.6293 301.3997 302.2071	6.2083 -0.0349 6.8191 -0.0349 7.6131 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.16106 -0.15332 -0.14623	-0.24660 -0.22896 -0.21217	-0.23166 -0.22500 -0.21904	-0.38981 -0.35817 -0.32535
413.5843 415.6733 418.1598	86.4199 86.4929 86.5797	303.0593 303.9575 304.8921	8.6267 -0.0349 9.9012 -0.0349 11.5528 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.13995 -0.13486 -0.13082	-0.19680 -0.18385 -0.17322	-0.21305 -0.20734 -0.20063 -0.19345	-0.29479 -0.26742 -0.24520
421.1924 424.9686 429.7618	86.4929 86.5797 86.6856 86.8175 86.1998 87.4719 87.8037	303.9575 304.8921 305.8381 306.7483 307.5202 308.1167 308.6300 309.5431 309.5431 309.8979 310.1622 310.3740 310.6270 291.5580	9.9012 -0.0349 11.5528 -0.0349 13.7291 -0.0349 16.3941 -0.0349 19.5421 -0.0349	0.9994	0.0000 0.0000 0.0000	-0.13486 -0.13082 -0.12809 -0.12571 -0.12318	-0.17322 -0.16543 -0.15807 -0.14888	-0.18349	-0.26742 -0.24520 -0.225907 -0.21598 -0.20592 -0.16977 -0.14903
435.9188 443.7087 453.2100	87.1998 87.4719 87.8037	308.1167 308.6300 309.1111	26.4140 -0.0349 28.9863 -0.0349	0.9994 0.9994 0.9994	0.0000	-0.11929 -0.11580 -0.11341	-0.13546 -0.12032 -0.10597	-0.17186 -0.171869 -0.120444 -0.120444 -0.11475 -0.111722 -0.0721759 -0.62194832 -0.4219832 -0.421659	-0.16977 -0.14905 -0.13133
464.2882 476.5780 489.5138 502.3777 513.5960	88 1998 88 1998 89 0716 89 19100 889 191100 85 15818 85 1782 885 1882 885 19480 885 19480 885 19480 885 19480 886 1948 886 1948 886 1948 886 1948	309.8979 310.1622	30.4696 -0.0349 30.7875 -0.0349 30.0206 -0.0349 28.2416 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0000 0.0000 0.0000 0.0000	-0.11309 -0.12069 -0.13378	-0.09566 -0.08610 -0.08585 -0.09209	-0.11475 -0.11176 -0.14222	-0.13133 -0.11306 -0.10315 -0.10119
513.5960 381.8565 384.3531	89.9124 84.9105 85.2310	310.6270 291.0110 290.5580	23.5306 -0.0349 14.0480 -0.1535	0.9994	0.0000 0.0284 -0.0015	-0.17941 -0.32725 -0.21093	-0.09209 -0.12296 -0.50980 -0.38172	-0.08747 -0.72183 -0.32759	-0.11084 -0.66841 -0.50787
386.9702 390.3600	85.4241 85.5885 85.7318	290 . 5580 290 . 2938 293 . 3855 296 . 1344 298 . 2787 299 . 8113	9.9903 -0.0533 16.8342 -0.0409	0.9985 0.9992 0.9994 0.9994	-0.0080 -0.0032	-0.34642 -0.29071 -0.26471	-0.50980 -0.38172 -0.53158 -0.49909 -0.48999	-0.61938 -0.42482 -0.45253	-0.65336
396.4663 398.3756 399.8469 401.0072 401.85124	85.8224 85.8888 85.9402	298.2787 299.8113 300.8134	10.0865 -0.0347 7.7032 -0.0349 6.0283 -0.0349	0.9994 0.9994 0.9994 0.9994	-0.0001 0.0001 0.0000 0.0000	-0.24118 -0.23230 -0.22706	-0.44450 -0.40766 -0.39418	-0 41456	-0.63085 -0.61983 -0.60576 -0.59447
401.0072 401.8512 402.5374 403.2720	85.9807 86.0102 86.0342 86.0598	300.8134 301.3616 301.6180 301.8628	4.7934 -0.0349 3.1273 -0.0350 3.4923 -0.0349	0.9994	0.0000 0.0000 0.0000	-0.11580 -0.11580 -0.11341 -0.11379 -0.11309 -0.12069 -0.21093 -0.221093 -0.22471 -0.223230 -0.22082 -0.22082 -0.21478 -0.21478 -0.21478 -0.21099 -0.20657 -0.201529	-0.44450 -0.40766 -0.39418 -0.38574 -0.37986 -0.379880	-0.40280 -0.39482 -0.37927 -0.36498	-0.57344
404.0538 404.8915	86.0871 86.1163	301.8628 302.2258 302.6722 303.1745		0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.21478 -0.21099 -0.20657	-0.36880 -0.36246 -0.35483 -0.34537 -0.33364	-0.35026 -0.33660 -0.32179 -0.30676 -0.29056	-0.56647 -0.55978 -0.55995 -0.53868 -0.52334
405.8004 406.8000	86.1481 86.1830	303.1745 303.7170 304.2939	4.3528 -0.0349 4.7026 -0.0350 5.1275 -0.0349	$0.9994 \\ 0.9994$	0.0000	-0.20135 -0.19529	-0.34537 -0.33364	-0.30676 -0.29056	-0.53868 -0.52334

770799378055719909859885983943867555504664894802235523699185686652158768987288598859483874188293675555046648942235567555046648897695758555555555555555555555555555555	145477476913691366905242115921669475575250057743445775250557453445679147669137456792166994755752500574169477577885555555555555555555555555555555	304567484481334747893307447850033333333333333333333333333333333333	5.3924 -0.0349 6.3324 -0.0349 7.31336 -0.0349 9.9490 -0.0349 11.8421 -0.0349 12.7533 -0.0349 12.7533 -0.0349 12.7533 -0.0349 12.7533 -0.0349 12.7533 -0.0349 12.8504 -0.0349 12.8504 -0.0349 12.8504 -0.0349 12.8504 -0.0349 12.8504 -0.0349 13.10166 -0.0349 14.7884 -0.0349 15.164 -0.0349 16.7165 -0.0349 17.884 -0.0349 17.884 -0.0349 18.5371 -0.0349 19.0412 -0.0349 19.0412 -0.0349 19.0412 -0.0349 19.0412 -0.0349 10.1444 -0.0349	44444444444444444444444444444444444444	0.0000	-0.18817 -0.18817 -0.1871768 -0.177168 -	0363359886848472235536331181224471810230568838024422384498908334693353463311812214747481023056883802442238449830753242131152136078681240213955092111666323391774747	4438144680544027959815576188810003992358882592449421973578912451292286289269892693892357888259924994219916439026579211884312992862892663889259347276660218433951831209285577000011111121342628926621855535110098844755494795677121520998514789267700000000000000000000000000000000000	-0.45275433405113784030000332513135986333251313598633336313135986333235131359863332351313598633323513135986333235131359863332335333591335913359133591335913359133
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### plate top middle

66211877588796433581999034111693346452146003366621621543880639498902751223886657498872091593856621855540667887964355554066578987209159385662169385348666574385555406214856555340639024751223886665747287555555555555555555555555555555555	55577533008820443258111088866149160266382554659290891272366028818902805828037863574273096463661143157577533008820043258111088866149160266382554659290891277236660288058280037866357742730964636661141111122223344568061615161111122223344565079125882622624947456596149129229049335680493447456591495975572144283992677450794944745657911272137155161616111122223334564774561916161611111222233344577078507949494745679112721371551616161111222233344576079114711616161111122223334457707850794949474567911471622357557211411112222333445770785079494947456791147161616111122223334457707850794949474567911471616161111222233344577078507949494745679114716161611112222333445770785079494947456791147161616111122223334457707850794949474567911471616161111222233344577078507949494745679114716222354457707850794949474567911471616161111222233445770785079494947456791147161616111122223344577078507949494745679114716161611111222233445770785079494947456791147161616111112222334457707850794949474567911471616161111122223344577078507949494745679114771616161111122223344577078507949494745679114771616161611111222233445770785079494947456791147716161616111112222334457807949474577078507949494745679114771616161611111222233445770785079494947456791147716161616111112222334457807949474745679114771616161611111112222333445770785079494745770785079494747456791147716161616111111111111111111111111	26138652663396162355466326691151682706602447714060775544298511736998661886763688599507340282669269151033311123726555544432203237255555444322337232372323723237232372	AREA 0.0.03355000000000000000000000000000000	44444444444444444444444444444444444444	EZ 0.00001 0.000012 -0.000012 -0.000002 -0.00002 -0.00002 -0.00002 -0.00002 -0.00002 -0.00002 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.000000 -0.0000000 -0.0000000 -0.000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.000000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.00000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000 -0.0000000000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0065341314105000000000000000000000000000000	640002639244107502585711266981110589461769870960316991353509974638244571376733083255911530730120423353 323398654975886160261413991309815105977764267913334479133398654203653176406615050878 222222777777626665554333334412519209765431535443322122222222222222222222222222	\$\\ \frac{2}{2}\) 2030655417768734467416883836125745920248497361672222814453534416697900005550988497362873245554982915174690500005545559884973655598849736555988497365559884973655598849736559884973655598555988497365555988497365555598849736555559884973655555988497365555578849787878878878878878878878878878878878878
413.6158 415.5297 417.9022 420.8770 424.6399 429.4201 435.4765	86.4230 86.4899 86.5726	252.3151 252.3151 252.3151 2552.69.324 2552.69.325 2550.912681 22550.912681 2245.49.315 2245.49.315 2245.49.315 2245.49.315 2245.49.315 2245.69.315 2245.37784	2.7871 -0.0350 3.8448 -0.0350 5.3501 -0.0349 7.5025 -0.0349 15.0346 -0.0349 15.0346 -0.0349 21.2994 -0.0349 29.8480 -0.0349 40.9873 -0.0349 69.1753 -0.0349 82.9320 -0.0349 92.8785 -0.0349 92.8785 -0.0349 93.4692 -0.0349 93.4692 -0.0349 0.0472 -0.0355 0.1584 -0.0356 0.2970 -0.0356 0.2970 -0.0353	0.9994 0.9994 0.9994 0.9994 0.9994	-0.0002 -0.0002 -0.0001 -0.0001 -0.0001 -0.0001 -0.0001	-0.09866	-0.15098 -0.14416 -0.13663 -0.12845 -0.11952 -0.10983	-0.2496473 -0.2496473 -0.2496112 -0.22439515004 -0.222359305 -0.22135935 -0.22135935 -0.22135935 -0.21937838 -0.21937838 -0.21937838 -0.22651034 -0.22651034 -0.225743 -0.225743	-0.17545 -0.16543 -0.15495 -0.14433 -0.13374 -0.11356

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434.7197 442.3747 451.8145	87.1589 87.4260 87.7555	269.9642 271.3729 272.7477	30.6616 -0.0349 41.8966 -0.0349 55.4547 -0.0349	0.9994 0.9994 0.9994	0.0001 0.0001 0.0000	-0.10255 -0.09692 -0.09218	-0.10244 -0.09387 -0.08507	-0.18472 -0.17585 -0.16857	-0.11832 -0.11085 -0.10261
463.1374 476.2347 490.7405	88.1509 88.6082 89.1148	273.9931 274.9964 275.6395	70.2128 -0.0349 83.9550 -0.0349 93.9094 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.08831 -0.08721 -0.08689	-0.07614 -0.06941 -0.06352	-0.16340 -0.16445 -0.16724	-0.09278 -0.08350 -0.07344
506.0407 521.4035 534.1325 405.3967	89.6490 90.1852 90.6296 86.1341	275.8452 275.5797 275.0354 257.3239	97.5373 -0.0349 94.2896 -0.0349 61.6825 -0.0349 0.0618 -0.0357	0.9994 0.9994 0.9994 0.9994	0.0000 0.0000 0.0000 0.0002	-0.09149 -0.10275 -0.08491 -0.17147	-0.06298 -0.06970 -0.05694 -0.19734	-0.18719 -0.21322 -0.23955 -0.27438	-0.06806 -0.06773 -0.05382 -0.24167
405.6904 406.0431 406.4735 407.0026	86.1445 86.1569 86.1720 86.1907	257.3239 257.9944 258.7074 259.4609 260.2578	0.2079 -0.0356 0.3890 -0.0354 0.6161 -0.0354 0.9046 -0.0354	0.9994 0.9994 0.9994 0.9994	0.0001 0.0001 0.0002 0.0002	-0.16615 -0.16130 -0.15744 -0.15356	-0.19375 -0.18740 -0.18108 -0.17497	-0.26584 -0.26080 -0.25575 -0.25078	-0.24270 -0.23645 -0.22837 -0.22028
407.6427 408.4174 409.3716	86.2131 86.2402 86.2736	261.1060 262.0025 262.9573	1.2603 -0.0352 1.7139 -0.0352 2.3489 -0.0352	0.9994 0.9994 0.9994	0.0002 0.0002 0.0002	-0.14970 -0.14585 -0.14192 -0.13786	-0.16872 -0.16244 -0.15609	-0.24593 -0.24099 -0.23610 -0.23086	-0.21158 -0.20243 -0.19274
410.5461 411.9991 413.8197 416.1196	86.3147 86.3655 86.4290 86.5093	263.9877 265.1003 266.3142 267.6436	4.3075 -0.0350 5.9139 -0.0350 8.1797 -0.0350	0.9994 0.9994 0.9994 0.9994	0.0002 0.0002 0.0002 0.0002	-0.13376 -0.12938 -0.12491	-0.14970 -0.14345 -0.13717 -0.13112	-0.22564 -0.21978 -0.21366	-0.18270 -0.17261 -0.16249 -0.15287
419.0502 422.8038 427.6148 433.7496	86.6116 86.7426 86.9105 87.1246	269.0985 270.6908 272.4176 274.2507	11.4197 -0.0349 16.0274 -0.0349 22.4702 -0.0349 31.2146 -0.0349 42.5548 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0002 0.0001 0.0001 0.0001	-0.12016 -0.11513 -0.10979 -0.10448	-0.12510 -0.11899 -0.11236 -0.10499	-0.20677 -0.19907 -0.19038 -0.18130	-0.14389 -0.13585 -0.12856 -0.12170
433.7496 441.4796 451.0289 462.4972 475.7744	87.1246 87.3945 87.7279 88.1283 88.5920	274 2507 276 1342 277 9744 279 6429 280 9878	42.5548 -0.0349 56.2458 -0.0349 71.1795 -0.0349 85.1289 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0001 0.0000 0.0000 0.0000	-0.09926 -0.09506 -0.09187 -0.09155	-0.09644 -0.08761 -0.07885 -0.07263	-0.17191 -0.16408 -0.15851 -0.15910	-0.11417 -0.10588 -0.09629 -0.08776
490.4889 506.0044 521.5632	89.1060 89.6478 90.1907	281.8517 282.1277 281.7696	98.8436 -0.0349 98.8436 -0.0349 95.4577 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0000 0.0000 0.0000	-0.09230 -0.09921 -0.11381 -0.11710	-0.06759 -0.06923 -0.07818 -0.08647	-0.16165 -0.18150 -0.20778 -0.23176	-0.07877 -0.07592 -0.07708 -0.08295
534.2579 405.3229 405.4676 405.6719	90.6340 86.1315 86.1365 86.1437	281.0444 257.3783 258.1643 259.0080	0.0675 -0.0351 0.2248 -0.0354 0.4176 -0.0354	0.9994 0.9994 0.9994	0.0001 0.0000 0.0001	-0.17275 -0.16809 -0.16378	-0.19643 -0.19516 -0.19076	-0.27278 -0.26418 -0.25861	-0.23668 -0.24174 -0.23994
405.9593 406.3521 406.8627 407.5208	86.1538 86.1676 86.1854 86.2084	259.9097 260.8734 261.9077 263.0128	0.6548 -0.0353 0.9509 -0.0353 1.3219 -0.0352 1.7915 -0.0352 2.4367 -0.0352	0.9994 0.9994 0.9994 0.9994	0.0001 0.0001 0.0002 0.0002	-0.16011 -0.15632 -0.15243 -0.14851	-0.18522 -0.17929 -0.17305 -0.16660	-0.25391 -0.24888 -0.24402 -0.23907	-0.23433 -0.22740 -0.21941 -0.21049
408.3716 409.4536 410.8288 412.5870	86.2084 86.2382 86.2760 86.3240 86.3854	264.1997 265.4866 266.8829 268.4110	2.4367 -0.0352 3.2730 -0.0352 4.4431 -0.0351 6.0860 -0.0350	0.9994 0.9994 0.9994 0.9994	0.0002 0.0002 0.0002 0.0002	-0.14442 -0.14017 -0.13591 -0.13138	-0.15993 -0.15318 -0.14662 -0.14011	-0.23404 -0.22863 -0.22322 -0.21713	-0.20058 -0.19008 -0.17937 -0.16858
414.8422 417.7480 421.4990	86.4641 86.5656 86.6966	270.0881 271.9263 273.9388 276.1213	8.4033 -0.0350 11.7087 -0.0350 16.4135 -0.0349 22.9944 -0.0349	0.9994 0.9994 0.9994 0.9994	0.0002 0.0002 0.0001 0.0001	-0.12675 -0.12192 -0.11691 -0.11166	-0.13392 -0.12795 -0.12204 -0.11563	-0.21072 -0.20350 -0.19547 -0.18638	-0.15830 -0.14887 -0.14058 -0.13311
426.3346 432.5270 440.3537 450.0435	86.8654 87.0816 87.3549 87.6932	278.4391 280.8211 283.1493	31.9220 -0.0349 43.4906 -0.0349 57.4384 -0.0349	0.9994 0.9994 0.9994	0.0001 0.0001 0.0001	-0.10651 -0.10160 -0.09774	-0.10839 -0.09985 -0.09092	-0.17680 -0.16693 -0.15854	-0.12598 -0.11810 -0.10944 -0.09938
461.6994 475.2107 490.1962 506.0009	88.1002 88.5721 89.0959 89.6477	285.2615 286.9646 288.0598 288.4095	72.6288 -0.0349 86.8003 -0.0349 97.0632 -0.0350 100.6786 -0.0348	0.9994 0.9994 0.9994 0.9994	0.0001 0.0000 0.0000 0.0000	-0.09496 -0.09520 -0.09685 -0.10701	-0.08205 -0.07580 -0.07114 -0.07533	-0.15261 -0.15290 -0.15610 -0.17641	-0.09009 -0.08066 -0.07994
521.8204 534.4435	90.1996 90.6404	287.9566 287.0508	96.9473 -0.0349 58.0400 -0.0349	0.9994 0.9994	0.0000	-0.12356 -0.14943	-0.08520 -0.10901	-0.20405 -0.22896	-0.08225 -0.10347

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418 . 6123 421 . 5998 425 . 3159 430 . 0384 436 . 1207 443 . 8425 453 . 2944 453 . 2944 466 . 6191 489 . 5445 502 . 3975 513 . 5968 380 . 0363	86. 5995 86. 69945 86. 89945 86. 9945 87. 2069 87. 2066 87. 8066 87. 8066 87. 8066 88. 62127 89. 5215 89. 9123 89. 9123 84. 6898	209.0673 208.1243 207.2227 206.4577 205.8625 204.8575 204.8575 204.0472 203.7508 203.5551 203.3157 231.4459 231.9528	16.2211 -0.0349 19.3375 -0.0349 22.9158 -0.0349 26.3223 -0.0349	0.9994 0.9994 0.9994	0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000	-0.15809 -0.13201 -0.109063 -0.09766 -0.06902 -0.06936 -0.06936	-0.23259 -0.20024 -0.163304 -0.10704 -0.08848 -0.07505 -0.073632 -0.073632	-0.63938 -0.62136 -0.61027 -0.60946 -0.61185 -0.61310 -0.57485	-0.12148 -0.09491 -0.07329 -0.05651 -0.05504

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\$\frac{421}{421}.1618\$\$\$ \$6.6845\$\$ \$250.8931\$\$ \$13.8309\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.118933\$\$ \$-0.16118\$\$ \$-0.2917\$\$ \$-0.1605\$\$ \$425.8995\$\$ \$86.8500\$\$ \$228.4752\$\$ \$16.7385\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.118933\$\$ \$-0.16185\$\$ \$-0.288055\$\$ \$-0.1616\$\$ \$431.7882\$\$ \$87.0556\$\$ \$226.2048\$\$ \$20.1825\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.10738\$\$ \$-0.12630\$\$ \$-0.282655\$\$ \$-0.1413\$\$ \$439.1583\$\$ \$87.3130\$\$ \$223.9305\$\$ \$23.96217\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.08461\$\$ \$-0.09266\$\$ \$-0.28076\$\$ \$-0.1061645\$\$ \$489.4594\$\$ \$87.6332\$\$ \$221.71559\$\$ \$27.5847\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.08461\$\$ \$-0.09266\$\$ \$-0.28076\$\$ \$-0.1061645\$\$ \$489.4594\$\$ \$88.0219\$\$ \$219.6752\$\$ \$30.5017\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.06911\$\$ \$-0.07590\$\$ \$-0.08068\$\$ \$-0.28076\$\$ \$-0.1061645\$\$ \$489.4892\$\$ \$88.0219\$\$ \$217.9700\$\$ \$32.4687\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.06911\$\$ \$-0.07033\$\$ \$-0.29991\$\$ \$-0.0755\$\$ \$472.4482\$\$ \$88.4756\$\$ \$217.9700\$\$ \$32.4687\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.06128\$\$ \$-0.05023\$\$ \$-0.05035\$\$ \$502.1983\$\$ \$89.5146\$\$ \$33.8997\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.06128\$\$ \$-0.05022\$\$ \$-0.05025\$\$ \$502.1983\$\$ \$89.5146\$\$ \$38.9810\$\$ \$216.1884\$\$ \$33.9472\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.06128\$\$ \$-0.05022\$\$ \$-0.05025\$\$ \$502.1983\$\$ \$89.5146\$\$ \$38.5997\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.07074\$\$ \$-0.06628\$\$ \$-0.30705\$\$ \$-0.05025\$\$ \$528.8827\$\$ \$90.4462\$\$ \$217.4072\$\$ \$17.5864\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.05502\$\$ \$-0.04466\$\$ \$-0.2222666\$\$ \$-0.0395\$\$ \$528.8827\$\$ \$90.4462\$\$ \$217.4072\$\$ \$17.5864\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.05502\$\$ \$-0.04466\$\$ \$-0.222566\$\$ \$-0.0395\$\$ \$528.8827\$\$ \$90.4462\$\$ \$217.4072\$\$ \$17.5864\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.05502\$\$ \$-0.04466\$\$ \$-0.222566\$\$ \$-0.225165\$\$ \$-0.0395\$\$ \$528.8827\$\$ \$90.4462\$\$ \$217.4072\$\$ \$17.5864\$\$ \$-0.0349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.255502\$\$ \$-0.04466\$\$ \$-0.225055\$\$ \$0.0077\$\$ \$37.0606\$\$ \$3.2028\$\$ \$3.7955\$\$ \$12.2366\$\$ \$-0.0395\$\$ \$39.0359\$\$ \$85.9192\$\$ \$25.7950\$\$ \$13.3842\$\$ \$-0.05349\$\$ \$0.9994\$\$ \$0.0000\$\$ \$-0.28663\$\$ \$-0.50171\$\$ \$-0.322967\$\$ \$-0.29350
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406.2876 407.5122	86.1651 86.2079	245.8997 244.2611	4.0062 -0.0349 4.7974 -0.0349	0.9994	0.0000	-0.17118 -0.16610	-0.22528 -0.21768	-0.29174 -0.29057	-0.27429 -0.26136
409.0685	86.2622	242.5150 240.6510	5.7875 -0.0349 7.0295 -0.0349	0.9994 0.9994	0.0000	-0.16021 -0.15353	-0.20843 -0.19765	-0.28879 -0.28670	-0.24649 -0.23003
411.0269 413.4819	86.3306 86.4163	238.6578	8.6272 -0.0349	0.9994	0.0000	-0.14600	-0.18534	-0.28388	-0.21230
416.5617 420.4374	86.5239 86.6592	236.5255 234.2493	10.6771 -0.0349 13.3146 <b>-</b> 0.0349	0.9994	0.0000	-0.13754 -0.12786	-0.17151 -0.15602	-0.28038 -0.27603	-0.19371 -0.17449
425.3265	86.8300	231.8417 229.3380	16.5780 -0.0349 20.4040 -0.0349	0.9994	0.0000	-0.11687 -0.10457	-0.13873 -0.11995	-0.27003 -0.26254	-0.15487 -0.13506
431.4881 439.2126 448.7622	87.0451 87.3149	226.8047	24.5559 -0.0349	0.9994	0.0000	-0.09184	-0.10088	-0.25585	-0.11590
448.7622 460.2676	87.6484 88.0502	224.3394 222.0891	28.6277 -0.0349 32.2047 -0.0349	0.9994 0.9994	0.0000	-0.08044 -0.07160	-0.08409 -0.07117	-0.25120 -0.25260	-0.09903 -0.08411
473.6299	88.5168 89.0357	220.2405 218.9775	35.0137 -0.0349 36.9829 -0.0349	0.9994	0.0000	-0.06530 -0.06073	-0.06114 -0.05252	-0.26002 -0.26690	-0.06992 -0.05610
488.4785 504.1596	89.5833	218.4517	37.8988 -0.0349	0.9994	0.0000	-0.06138	-0.04838	-0.27018	-0.04817
519.6584 531.4355	90.1242 90.5354	218.7879 219.9119	35.9941 <b>-</b> 0.0349 18.0849 <b>-</b> 0.0349	0.9994 0.9994	0.0000	-0.07420 -0.09403	-0.05431 -0.07289	-0.29419 -0.25359	-0.04924 -0.06360
367.8432 372.9255	81.6496 83.2187	255.8415 255.8890	10.9461 -0.3580 13.0605 -0.2433	0.9337 0.9697	-0.0108 -0.0211	0.23515 -0.21291	0.17114	-0.20576 -0.43800	0.03744 -0.30910
379.0689	84.4315	255.9123	14.2609 -0.1487	0.9887	-0.0200	-0.39880	-0.21537 -0.47124	-0.49619	-0.52875
385.5711 391.5752	85.2161 85.6286	255.9283 255.9595	13.7163 -0.0901 11.3328 -0.0433	0.9959 0.9991	-0.0113 -0.0033	-0.40696 -0.36439	-0.56589 -0.60050	-0.45078 -0.42398	-0.62692 -0.68670
396.4664 400.0421	85.8225 85.9470	256.0278 256.1273	7.9745 -0.0347 4.8951 -0.0349	0.9994	0.0004	-0.26563 -0.20331	-0.46181 -0.26961	-0.34393 -0.30566	-0.62723 -0.42718
402.4377	86.0307	256.2031	2.8861 <b>-</b> 0.0349	0.9994	0.0000	-0.19428	-0.24307	-0.29662	-0.29402
403.9690 404.6308	86.0842 86.1073	256.1925 255.8042	2.0029 -0.0349 1.1344 -0.0349	0.9994 0.9994	0.0000	-0.18051 -0.17752	-0.21790 -0.21497	-0.28225 -0.28287	-0.28351 -0.26704
404.7483	86.1114 86.1156	255.0403 254.1696	1.1218 -0.0349 1.2874 -0.0349	0.9994	0.0000	-0.17596 -0.17423	-0.21577 -0.21565	-0.27711 -0.27775	-0.27235 -0.27189
404.8681 405.0491	86.1219	253.1976	1.5852 -0.0348	0.9994	0.0000	-0.17309	-0.21612	-0.27705	-0.27167
405.3367 405.7609	86.1319 86.1467	252.1282 250.9567	1.9685 <b>-</b> 0.0349 2.4169 <b>-</b> 0.0349	0.9994 0.9994	0.0000	-0.17158 -0.16973	-0.21589 -0.21486	-0.27753 -0.27803	-0.27016 -0.26699
406.3605 407.1782	86.1677 86.1962	249.6859 248.3125	2.9339 -0.0349 3.5493 -0.0349	0.9994 0.9994	0.0000	-0.16725 -0.16403	-0.21248 -0.20852	-0.27846 -0.27858	-0.26166 -0.25400
408.2507	86.2337	246.8238	4.2505 -0.0349	0.9994	0.0000	-0.16017 -0.15556	-0.20319 -0.19628	-0.27858 -0.27855 -0.27804	-0.24443 -0.23288
409.6390 411.4276	86.2822 86.3446	245.2127 243.4601	5.1471 -0.0349 6.2897 -0.0349	0.9994 0.9994	0.0000	-0.15022	-0.18788	-0.27726	-0.21960
413.7262 416.6812	86.4248 86.5280	241.5493 239.4648	7.7822 -0.0349 9.7351 -0.0349	0.9994 0.9994	0.0000	-0.14397 -0.13674	-0.17780 -0.16604	-0.27581 -0.27356	-0.20463 -0.18837
420.4837 425.3721	86.6608 86.8316	237.1968 234.7553	12.3032 <b>-</b> 0.0349 15.5846 <b>-</b> 0.0349	0.9994	0.0000	-0.12828 -0.11840	-0.15247 -0.13694	-0.27044 -0.26596	-0.17113 -0.15327
431.6144	87.0495	232.1764	19.6123 <b>-</b> 0.0349	0.9994	0.0000	-0.10704 -0.09505	-0.11975 -0.10218	-0.26009 -0.25540	-0.13513 -0.11755
439.4891 449.2307	87.3245 87.6647	229.5380 226.9615	24.3025 -0.0349 29.4123 -0.0349	0.9994 0.9994	0.0000	-0.08415	-0.08690	-0.25318	-0.10198
460.9488 474.5375	88.0739 88.5485	224.6212 222.7259	34.5760 -0.0349 39.3070 -0.0349	0.9994 0.9994	0.0000	-0.07545 -0.06907	-0.07510 -0.06551	-0.25649 -0.26533 -0.27269	-0.08763 -0.07322
489.6174 505.5287	89.0756 89.6312	221.4780 221.0391	43.0957 -0.0350 45.2895 -0.0348	0.9994	0.0000	-0.06397 -0.06390	-0.05633 -0.05117	-0.27269 -0.27587	-0.05867 -0.04958
521.3676	90.1838	221.4954 222.5693	44.2677 -0.0349 22.2800 -0.0349	0.9994	0.0000	-0.07528 -0.06424	-0.05601 -0.05007	-0.29127 -0.23223	-0.04967 -0.03942
533.6021	90.6110	222.3093	22.2000 -0.0343	0.5554	0.0000	0.00727	0.00007	0.20220	0.00011